

April 20, 1929

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*The Oldest American Aeronautical Magazine*



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High Speed	175 mph
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DESIGNED from experience gained flying 7,000 miles daily over the longest airplane laboratory in the world! Model 95 proves itself in the toughest among mail-cargo planes by virtue of SPEED, HEAVY PAY LOAD, (1,610 pounds), QUICK TAKE-OFF, LOW LANDING SPEED, ECONOMICAL OPERATION.

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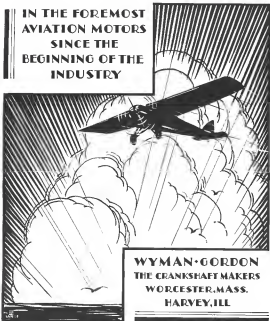
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WE PLACE at your command one of the world's most complete funds of knowledge about hangars and their construction.

Robertson engineers have been taking part in hangar construction since the beginnings of modern commercial aviation. They have been all over the world, learning invaluable lessons, witnessing and often participating in most of the notable experiments that have been made in methods of building and equipping hangars. They know, from actual experience, the things that will work, and the things that won't.

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The Robertson engineers have prepared a booklet which gives the reasons for and explains various building materials, general information regarding possible costs, and illustrations of many types of hangars. It is a clear statement of the world's best 12 years of experience with hangars. It will be of great value to anyone planning a hangar. It will be sent without charge.



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THEY KNOW FOR MONTHS IN AVIATION

## AVIATION

THE OLDEST AMERICAN AERONAUTICAL MAGAZINE

April 30, 1939

VOLUME 11 NUMBER 11



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## Pooling Patents

COMMERCIAL manufacturers of airplanes have not bothered much about patents. The truth is, thus far, they have not been enough commercial patents to make it worth while for the owners of patents to do any thing. Commercial manufacturers have thus been lulled into the belief that no one would bother them about patents and most of them do not know what patents, if any, they are infringing.

Among the suppliers to the Government, the situation was different and, to avoid disastrous patent litigation, during the War, patents were pooled under an association called the Manufacturers' Aircraft Association.

Since the War, the Association has continued to function and, on the whole, the patent situation has worked out more satisfactorily than in most new industries. The general manager of the Association has been S. B. Bradley, and it is due largely to his quiet diplomacy that matters have run so smoothly. Recently, the cross-licensing agreement has been revised, and under its new form, the license cost per plane has been lowered and put on a sliding scale. The agreement pooling, as it does, some three hundred of the most important patents, from the members from four of suits for infringement. It also makes a provision for the payment of new patents at a price to be set by a board of arbitration. If manufacturers of commercial planes are to avoid themselves of the inventions of others and avoid disastrous patent suits, they must become part of some such association. There is nothing mandatory about joining the Manufacturers' Aircraft Association, but the Association has done fine work for military manufacturers and it is probable that the commercial manufacturers will find it expedient to join the Association.

## Speed or Regularity

RAILROADS many years ago reached what seems to be the limit of speed from the practical operating standpoint. With extra travel there is an absolute definite limit beyond which any increase in speed is only achieved by a tremendous increase in cost. The airplane alone has not yet reached the point where increased speed is practically impossible. As fast there are almost limitless possibilities of increased speed with little increase in danger, especially when we consider that profits are decreased, not by the hours flown but by the miles covered during the hour.

It would seem then that next to safety the only objective of an airline operator should be the achievement of greater speed. This is essentially true, but on the other hand, the operator must be sure that his schedule is such as to allow for the minor delays which necessarily must arise. A reasonable percentage of the increase in the speed which is being obtained with modern planes should be reserved for making up lost time, for badging head winds, and for throwing the engine under normal conditions.

With the increasing network of airlines it is becoming more important to keep up to schedules in order to make connections. Airlines must every other vehicle as speed, but they still lag in the regularity of their scheduled operations. In spite of the temptation to the contrary it would seem inadvisable to increase the speed of the schedules at a proportionate rate with the increase of the speed of the planes.

//

## High Speed

THE high speed record for land travel of over 230 m.p.h. which was established by Major General in Dayton was made possible through the research work done for aeronautical purposes. Both the streamlining of the automobile and the light weight (power produced) by the Napier engine are the results of aeronautical research experimentation. Aviation has thus begun to repay its part some of the debt it owes to the automobile, and those who have engineered automobile engines. Accord-

# THE *Detroit Show* FROM A Technical Angle

THE SECOND ANNUAL Air America Aircraft show was particularly valuable from an engineering standpoint, in that it afforded an opportunity to study some of the best airplane designs as well as the details of a large number of engines and accessories. Such a representation comes as nearly as possible to presenting an accurate picture of the status of aeronautical engineering.

Development in aircraft design, like other forms of progress, is a process of evolution. It is always possible, however, when a large number of airplanes are displayed, to note general design tendencies and to visualize the airplane of the future.

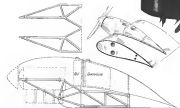
A comparison with the first All American Aircraft show, held a year ago in the same city, shows a greater proportion of airplanes of proven commercial value. It is interesting to note, however, that several of the

The importance of aerodynamic elements in obtaining increased performance is being realized where, a year ago, engineers would invariably increase power to gain the same end. Many creditable attempts have been



A drawing of the tail section of the Lockheed "Vampire" shows the "Vampire" power plant installed in a landing streamer fixed into the wing.

Sketches showing the method of adjustment of angles according to wing position. The position and size of the wing are shown in the sketch.



designs then regarded as speculative, have taken their place among the accepted types.

An attempt to simplify designs and mechanisms with the ultimate idea of high quantity production was observable on one side and several of the features which cause production "bottle necks" have been eliminated. On the other hand, there were a number of designs in which aerodynamic efficiency was considered before production possibilities.

In almost all of the last mentioned cases, however, it is probable that production methods will be improved and volume increased to the point where manufacturing costs on highly efficient designs will not be prohibitive when a large number of units are to be built.

made to reduce parasite drag and interference effects on the new models exhibited at Detroit.

One of the most significant steps in this direction is the adoption of the NACA low drag radial engine covering to airplanes and four of these were exhibited at the show. Increases of top speed from seven to ten m.p.h. and some greater were reported by the manufacturers whose planes embodied this principle. In most cases the cowling was made to follow, hinged together in such a manner that it could be removed without taking off the propeller. The cowlings exhibited varied widely in manner design. An interesting and useful modification of the cowling was found on the Pitcairn PA-6 biplane. This consists of a telescoped

section at the rear arranged to slide longitudinally and change the size of the annular opening between fuselage and cowling. The sliding section may be moved to any desired position and is held in place by several bolts with wing nuts.

This arrangement was adopted by the Pitcairn company to provide adequate cooling for warm weather operation. An additional step in this direction would be provision of a means for controlling the device from the pilot's cockpit.

Another efficient cooling installation was found on



A drawing showing the telescoped section of the NACA type of cowling installed on the Pitcairn PA-6.

the "Chaucer" biplane, exhibited by the Paramount Aircraft Corp. In appearance this plane is particularly well adapted to the cooling installation. This together with the special cooling process developed by the company, produced one of the most attractive exhibits of the show. The cowling was so hinged into the lines of the fuselage that it greatly improved the appearance of the plane and removed the machinery look that so attracts itself on the eye in the average radial engined plane.



Front quarter view of the "Chaucer" biplane with the NACA-type of radial engine cooling.

Tapered wings are coming into more general use and more originality is being shown in the development and choice of wing sections. Judging from the planes exhibited at Detroit, cantilever wing types are becoming popular and thicker wing sections are being used, providing greater spar depth and structural strength.

Although wood is still the predominant structural material in airplane wings, several of the new planes at the show had metal wing structures and two made use of both steel and aluminum alloy. Another craft exhibited had the structure of the wing. These will be described in greater detail later. Several conventional wood and metal wing structures also were employed.

For manufacturers using space as structural material in wings, an interesting and efficient arrangement has been worked out by the Peery Manufacturing Company. By this arrangement, the Peery Company contracts with the airplane manufacturer to furnish complete sets of spars, ribs or both, finished to the specifications of the aircraft manufacturer and ready for assembly. This eliminates not only waste of raw material but uncertainty regarding the proportion of suitable airplane spars in any given airplane, which is sometimes widely variable. A number of leading manufacturers are taking advantage of this service. Several of the finished products were exhibited at the Hand Starter in Detroit during the show.

An interesting metal wing structure was noted in the PT-66 motor biplane developed and exhibited by the Cunningham-Hall Aircraft Corporation, of Rochester, N. Y. This plane is one of the type having upper wings mounted flush with the top of fuselage, has a gross weight of 4,000 lb. and is powered by the 300-hp. Wright J-6 engine. Wing ribs are of Warren truss type of 1 in. O.D. 128 in. wall aluminum tubing, heat treated, with special metal joints. The upper wing houses are of one piece type, comprising upper and lower longitudinal of 1 1/2 in. O.D. by 5/16 in. wall chrome molybdenum steel tubing, heat treated to 125,000 lb. per sq. in. The upper longitudinal

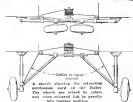
tudinal is reinforced for a considerable distance at the outer support by a 1½-in. O.D. 988 in wall chrome molybdenum steel tubing (flexible heat treated). The longitudinals are spaced by double vertical and diagonal members. These members are bolted to the longitudinals by 5/16 in bolts and 7/16 in spacer tubes. Drag trusses in the planes of both upper and lower longitudinals are used and compression struts are ¼ in. 357 in wall heat treated duralumin tube Warren trusses. Aileron

Among the interesting new designs seen at Detroit was the Eaglehawk "Bulfinch," designed by the Alcoa under duPont Corp. The Bulfinch is a four place, low wing, full cantilever cabin monoplane with landing gear

A side view of the Eaglehawk "Bulfinch" powered with a 165 hp Wright J-6 engine. This plane was exhibited for the first time at Detroit.



The advanced wing structure of the Bulfinch showing the internal members and internal bracing.



brings are located at compression strut points and aileron structures are of No. 1025 carbon steel.

An unusual production feature is found in the fuselage construction of this airplane. Longons are spaced at the joints by means of a punch press, thereby avoiding the usual curved level joints of the sheeting members and greatly facilitating production. The PT-6 will be described in greater detail in an early issue of Aviation.

The second example of duralumin and steel wing structure was that of the Cirrus (powered) British Blackburn "Bluebird," biplane which is soon to be produced in this country. This construction consists essentially of steel spars and duralumin ribs and fairing. Ribs and bolts are used throughout and there are no welds in the primary structure. This is also true of the fuselage structure, which is steel tubing with duralumin fairing members.

A duralumin wing structure is embodied in the design of the LeBlond (powered) Birking NB-3 low wing monoplane, developed by the Nicholas Birking Aeroplane Co. This structure consists mainly of a large duralumin box member. Bifurcated wing tips in another feature of this interesting design. Several plans having tapered aluminum alloy ribs with wooden spars also were exhibited.

The landing gear mechanism is simple and consists essentially of a drum and cables that draw the wheels up over streamlined components in the under side of the fuselage. By releasing a trigger the landing gear can be dropped into position, falling by gravity. It is kept in landing position by a double safety lock. A hand wheel mounted on the front wing beam is used to retract the landing gear.

The tail sled is a five leaf spring with an easily detachable hardened steel shoe. The entire assembly is pivoted to the fuselage and can be removed easily when necessary. The rudder area is set in such a position as to be vertical when the plane is taking on the ground or in a stall. Entrance doors are set in the fuselage at such an angle as to be horizontal when the plane is on the ground. The stabilizer is adjustable by a triple screw thread.

Another interesting design which made its first appearance at the Detroit Show was the "Ranger" exhibited by the Mohawk Aircraft Corp. The Ranger is a three place, low wing, cabin monoplane of the full cantilever type. It is powered by two four cylinder in line air cooled "Rover" engines mounted in housings

streamlined into the upper surfaces of the 64-ft. outer wing panels. These structures are built up of welded steel tubing, the longitudinals of the forward wings being 2 in. by 105 steel. The welded steel tube engine mounting is attached to the wing structure by not bolts and four bolts are employed to mount each inverted engine. Gasoline and oil tanks are located at the stream-lined engine housings. Use of other power plants is made possible by the detachable engine mounting.

The Ranger has a wing span of 46 ft., a length of 26 ft., a gross weight of 2000 lb. and a useful load of 600 lb. The tapered wing is of the internally braced full cantilever type. A split type landing gear and a tail wheel are provided and both the main and auxiliary are adjustable in flight.

Cabin highbacks having upper wings supported on struts above the fuselage were lower than at the Chicago Show but our new craft of that type was exhibited by the Knoll Aircraft Corp. The Knoll KN-1 has a wing span of 33 ft. 6 in., an overall length of 22 ft. 3 in., a gross weight of 3025 lb. and a capacity of six persons. It is powered by a Wright Whirlwind J-5 engine but also will be available with the 300-hp. J-6 power plant.

The KN-1 has a conventional fuselage structure and a wooden wing with plywood covering. An innovation is found in the methods of mounting engine instruments and controls. The engine instruments are placed on two small panels which are attached to the rear member of the engine mounting. Controls for the power plant also are attached to the mounting, which is of the four post type and hinges to the fuselage in such a manner as to swing downward when repairs or adjustments are to be made to the engine. When engine and mounting



A view through the uncompleted fuselage structure of a Trans-Atlantic Model PT-6 with 165 hp engine.



A sketch showing the internal structure of the fuselage of the PT-6.

retractable in flight, and is powered with a 100 hp Rover K-5 engine. The plane has a wing span of 34 ft. 7 in., an overall length of 21 ft. 7 in., a weight empty of 1150 lb. and a gross weight of 2250 lb. It is also available with the 165-hp Wright J-6 engine.

The Eaglehawk Bulfinch represents a step in the direction of the "flying wing" type of craft which is now being developed abroad. The wing consists of three units, i.e., two panels and a center section. Contrary to conventional practice, most of the support loads are borne by the steel tube center section of the wing. The fuselage carries engine and tail loads and provides a streamlined housing for the passengers. All wood has been spars and ribs having ribs in clear only are employed in the wing panel structure.

A specially derived airfoil section is used and a long elliptical taper reduces induced drag.

The wing panel structure of the PT-6 which is built up of steel tube spars and duralumin ribs.





A front quarter view of the new Whittaker-powered Beech A-1-C, the latest cabin biplane.

are in position the two instrument panels fit into cutouts in the instrument board which is mounted in the control compartment and contains the flight instruments. The control handles pass through a cut in the fire wall and are in the center of the space below the instrument board when the engine mounting is in position. By this arrangement it is possible to connect all engine instruments and controls before the engine is placed in the plane and much time is saved in changing engines.

That tail wheels are increasing in popularity was clearly shown at the exposition and a number of manufacturers who offer brakes as optional equipment are also offering tail wheels with the brakes. A novel tail roller of almost spherical shape was installed on the "Cavalier" plane at the show. Sprockets adjustable stabilizers are becoming more common and many are actuated by cables or combinations of cables and push pull tubes.

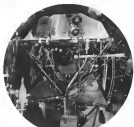
A consolidation shield and tail wheel was shown as the Partridge 71, the shield being of the large broad sheet type with a small tail wheel mounted on the shield about in such a way that the plane can be easily rolled about in concrete board hangars without the use of a dolly.

Several original production practices are employed in the manufacture of the biplane exhibited by Fleet Aircraft Inc., one of the more interesting being the use of Parler-Kalco brand aerial screws to attach the fabric to the stressed aluminum ribs.

**R**EFINEMENT that will place aircraft at the same level as automobiles as far as public acceptance is concerned was evident in a line of the planes exhibited. The new Versatile Air Coach, particularly, has been wisely cleaned up and has no trailing members to mar cabin lines as viewed from windows. This plane is well streamlined even to the degree of carrying the fuselage lines back into the lower portion of the cowl, and is finished so tastefully and expensively as the finest automobile.

That some order may materialize from the chaos of color schemes, which have heretofore been tried on various planes, was indicated by the Berry Brothers display with colored patches of many popular airplanes. One of the planes in the exhibit of the Hamilton Company had a Streamlined exterior finish.

An innovation in landing lights was shown by the



A close view of the engine mounting of the Beech plane showing fuel system and control mechanism.

Pittsburgh Company and consists of an aluminum lined compartment built into the leading edge of the wing with a window of glass curved to conform with the leading edge. The lens is mounted with a reflector only, the combination used being such that there is no need of a lens. The result is a perfectly streamlined and well protected leading light which should prove to be an efficient unit.

A steady trend toward the monoplane is new designs is indicated by the showing of two more monoplane than biplanes this year whereas there were considerably more biplanes than monoplane shown last year. The latest designs bear out the belief of engineers that there will always be a field for both the monoplane and the biplane with the former trend probably indicated by the large number of light sport and training planes of lightness type, while the biplanes are monoplane. The steps being taken by designers are further indicated in the presence at the show of seven folding wing airplanes, six low wing planes, three center wing planes and fifteen planes with wings of variable construction without any external bracing.

The open place led the closed in number of designs

exhibited. Despite the great opportunity for seaplanes and flying boats, there were none amphibious and seaplanes shown this year thus far. This may have been due, however, to the limited space, as at least two water craft were in the Detroit River during the show and could have been exhibited if space permitted. The Eisele Flying Boat appears to be a marked improvement over last year's model.

Of the light tri-engine class, the Kestrel Air Coach, powered with three LeRoi engines, was the only representative. It is possible, however, that there will be more light multi-engine planes in the future.

A considerable shift is evident in the seating capacity of the planes shown. This is particularly noticeable because of the fact that three times as many two place planes were shown this year than in 1936. This indicates an effort to develop good sport and training planes to fill the need for craft upon which the great number of pilots now desiring experience may be trained.

The three place open planes have not advanced in popularity, a lower percentage having been exhibited this year, but the three place closed plane seems to have gained favor for many monoplane of this type were shown. There is evident a tendency toward the four place plane providing a closer cabin interior than does that of the usual arrangement of a large number of seats. Five place planes have decreased in numbers and six place has increased greatly in numbers shown. An increased number of transport planes having capacities of 8, 10 and 12 persons was exhibited this year.

Planes equipped with radial engines have almost doubled while those using the OX-5 have decreased more than 25 per cent. This is partly due to the scarcity of OX-5 engines. The engine problem in the 1,000-2,000 h. gross weight class has been met by the redesign of

coolings for this type of engine have been developed.

Planes in the 2,000-2,500 h. gross weight range predominated in numbers and the increasing importance of the two place tandem sport and training types was very evident. Two of these craft had single cockpit cutouts, the persons being in tandem orientation between pilot and student. In most of the low wing planes, greater protection is provided for the occupants.

Among the engines exhibited, one of the outstanding tendencies seemed to be the direction of supercharging, and built-in superchargers were exhibited in many engines in the medium horsepower ranges of radial types.



A sketch of the new type of turbo-supercharger with a section of the intake and outlet pipes.

Among the engines shown there were but three under 50 hp. but the majority of the remainder were in either the 50-100 hp. or above 100 hp. classes, indicating that designers are attempting to develop more of the very light and very heavy engines, the medium class being fairly well developed at the present time.

**A**N AIR-COOLING "tree" type engine was exhibited by the American Aeronautical Corporation. This power plant develops 420 hp. with a dry weight of only 600 lb. Another Pratt & Whitney engine of particular interest was the type 80-1, a six cylinder in line aerobically designed weighing 240 lb. and developing 80 hp. The power plant has the entire upper portion of the crankcase and all cylinders and fuel inlet in one piece with a separate head bolted to each cylinder. The fins are thick and set far apart giving the engine a novel appearance in comparison with others of its type.

The four cylinder in line air-cooled type was represented by the Wright Gyro, the Mohrman Series Company "Rover," American Canard and Dupont "Blue" engines. One two cycle engine of novel design was exhibited by the Harrison Motors Corp. The new "Wasp" engine which was exhibited by the Pratt & Whitney Aircraft Corporation.

A "nose-thrust" lubrication system for radial engines was installed on the Lycoming Model R-645 radial engine exhibited at the show. This system is controlled from the cockpit and each rocker box supplied with oil lines running through the streamlined push rod housings.

Among the features of the new Continental radial engine are a special bearing for the counterbalance and a steel locked inhibitor rod bearing. The lubrication system is such that there are no oil lines.

Included in the power plant exhibits was a propeller of unique design. This propeller, displayed by the Pittsburgh Screw & Bolt Corporation, was hollow, built up to two flat sections of S-AE 6130 chrome vanadium steel welded together and unusually well finished. This propeller was used recently by Capt. Conrad C. E. Hines, U.S. Air Corps, of the Wright Aeronautical Corporation at Dayton, N. J.

A comparative table showing the types exhibited in 1936 and 1938.

	1936	1938
Total planes exhibited in Detroit Show	140	150
Monoplane aircraft	100	110
Biplane aircraft	40	40
Comparison between 1936 and 1938		
Single place	10	10
Two place	20	20
Three place	10	10
Four place	10	10
Five place	10	10
Six place	10	10
Seven place	10	10
Eight place	10	10
Nine place	10	10
Ten place	10	10
Eleven place	10	10
Twelve place	10	10
Thirteen place	10	10
Fourteen place	10	10
Fifteen place	10	10
Sixteen place	10	10
Seventeen place	10	10
Eighteen place	10	10
Nineteen place	10	10
Twenty place	10	10
Twenty one place	10	10
Twenty two place	10	10
Twenty three place	10	10
Twenty four place	10	10
Twenty five place	10	10
Twenty six place	10	10
Twenty seven place	10	10
Twenty eight place	10	10
Twenty nine place	10	10
Thirty place	10	10
Thirty one place	10	10
Thirty two place	10	10
Thirty three place	10	10
Thirty four place	10	10
Thirty five place	10	10
Thirty six place	10	10
Thirty seven place	10	10
Thirty eight place	10	10
Thirty nine place	10	10
Forty place	10	10
Forty one place	10	10
Forty two place	10	10
Forty three place	10	10
Forty four place	10	10
Forty five place	10	10
Forty six place	10	10
Forty seven place	10	10
Forty eight place	10	10
Forty nine place	10	10
Fifty place	10	10
Fifty one place	10	10
Fifty two place	10	10
Fifty three place	10	10
Fifty four place	10	10
Fifty five place	10	10
Fifty six place	10	10
Fifty seven place	10	10
Fifty eight place	10	10
Fifty nine place	10	10
Sixty place	10	10
Sixty one place	10	10
Sixty two place	10	10
Sixty three place	10	10
Sixty four place	10	10
Sixty five place	10	10
Sixty six place	10	10
Sixty seven place	10	10
Sixty eight place	10	10
Sixty nine place	10	10
Seventy place	10	10
Seventy one place	10	10
Seventy two place	10	10
Seventy three place	10	10
Seventy four place	10	10
Seventy five place	10	10
Seventy six place	10	10
Seventy seven place	10	10
Seventy eight place	10	10
Seventy nine place	10	10
Eighty place	10	10
Eighty one place	10	10
Eighty two place	10	10
Eighty three place	10	10
Eighty four place	10	10
Eighty five place	10	10
Eighty six place	10	10
Eighty seven place	10	10
Eighty eight place	10	10
Eighty nine place	10	10
Ninety place	10	10
Ninety one place	10	10
Ninety two place	10	10
Ninety three place	10	10
Ninety four place	10	10
Ninety five place	10	10
Ninety six place	10	10
Ninety seven place	10	10
Ninety eight place	10	10
Ninety nine place	10	10
Hundred place	10	10

30-35 hp and by the four cylinder in line air cooled engines. Ten planes powered with in line air cooled engines were shown where there were no instances of this type last year. Several very clean and attractive

# LOOKING BACK AT THE

## All-American Aircraft Show

By R. SIDNEY BOWEN, JR.

THE second annual All-American Aircraft Show, recently held in Convention Hall, Detroit, was the largest presentation of aircraft, engines, and equipment ever made in the history of the aeronautical industry, yet its attraction to the public at large, as judged by the attendance, was not on a par with the attraction of the 1926 show. A total of 193 planes was exhibited and of that number 49 were land monoplane, 47 were land biplanes, one was a triplane, one was a triplane amphibian, one was a flying boat, and one was a regular amphibian.

The individual exhibits, as regards attractiveness of display, showed very little improvement over last year. This, however, was not due entirely to the exhibitors. So great was the number of planes in exhibition that they had to be haphazardly dovetailed together that making it somewhat difficult for a visitor at the Show to walk about a particular plane and get a worthwhile view from all angles. There were, of course, a few exceptions to this condition. The outstanding of these was the exhibit of Curtiss Flying Service, Inc. which showed a Curtiss "Flying" training plane, a Curtiss (Challenger) "Robin," a Curtiss (D-53) Robin, and two four-place Curtiss monoplanes, together with biplane trainers, etc. The layout of the Curtiss Flying Service display was efficient, attractive and most effective. Curtiss training covered the entire floor space, which was surrounded by a white white picket fence, and chairs in which the weary visitor could rest and get his breath were placed at strategic points about the display. That, incidentally, was an idea that was used extensively by its exhibitor at the Show. With things so "jammed" together, as it were, visitors had no place to rest and were forced to keep wandering about on the second floor of the hall which dealt some too glibly with one's feet.

In fact, it is believed that the absence of pilots to meet around many visitors to leave before they had viewed all of the Show. Had there been some casual spot provided with chairs, and perhaps an architect to make a bit of special arrangement, it is almost certain that the Detroit Show would have made a far more favorable impression upon the visiting public and the visiting trade. Two other examples of attractive and conservatively specimen display were the exhibits of the Great Lakes Aircraft Co., and the Chance Vought Company. The former had one of its Curtiss powered training planes mounted in flying position upon a revolving platform. This one could stand still and inspect the plane as it

slowly revolved. The latter was similarly set off by the fact that it was a seaplane amphibian with silver and blue trimmings and its metal parts highly polished, with the exception of the float which was painted a Macgregor Blue. Displays were roped off by blue felt covered ropes which gave them the appearance of being separate and apart from the rest of the exhibits.

There were several other exhibits roped off as a similar manner, and although such an arrangement added a bit to the attractiveness of the display, it had its dis-



A view of the Westinghouse Electric Mfg. Co. booth at the Westinghouse Electric Co. booth at the right.

advantages as judged by watching the spectators roam about the Hall. The outstanding one was, that, to the best of the exhibit, it served somewhat as a "keep-off-the-grass" sign, with the result that the spectator viewed at a distance that which he would undoubtedly have liked to inspect closely. The suggestion is made that exhibitors in future aircraft shows arrange their displays so that the restricted parties may have the opportunity to "see for themselves."

In the case of automobile shows, the visitor is allowed, in that event, to get into the car and see how the seat fits, try the gear shift, the clutch, the brake and various other things. However, at aircraft shows, the visitor has little chance to sit in a plane and manipulate the "steer" and rudder pedals as he desires. It is admitted that a plane is not an automobile and that a certain amount of damage can be rendered to the former by over-zealous spectators. Yet, on the other hand, as was clearly demonstrated at the Detroit Show, the pub-

lic's interest in just looking at airplanes has waned considerably. Therefore, the exhibitor should make provisions whereby the spectator can get into the plane on exhibit and actually work the controls for himself. It may seem that those in charge of the exhibit will have to be more on the alert, but they should be.

As a matter of fact, exhibitors anxious to utilize was most conspicuous by its absence at Detroit. At an area show one can hardly look at a car before some salesman steps up and asks if he may be of service in explaining about a particular car. Whereas at Detroit, one could virtually stare at a plane until he was blue in the face before anyone in charge of the exhibit would offer any information. When the visitor at the Detroit Show wanted information he had to ask for it.

Perhaps that condition is partly explained by the fact that in many cases those left in charge of the plane displays did not know a great deal about their own product. In just those the construction of plane components were to be seen at their exhibits. At the Detroit Show they

were either attending the car business and one different meetings and trade conferences at the various Detroit hotels, or else at the Ford Airport building and watching on competitive planes as they took off and landed. In short, the exhibitor management of aircraft shows seems to have passed into the hands of amateurs and novices yet rather uneducated young men. In other words, the second annual All-American Aircraft Show was really eight days of trade conventions with an aircraft show and as a result of concentrating the activities of the industry at one point for a certain period of time. It has been estimated that ten times as much business was conducted behind closed doors in the three trading hotels of Detroit than was conducted inside of Convention Hall.

THAT SUCH AN ASSORTMENT of business should be conducted outside is altogether satisfactory, but the aircraft and engine manufacturers should realize that it is the public that ultimately buys airplanes and engines. When a prospective customer visits an aircraft show, asks a question of some shop in charge of an exhibit, and receives the reply that "...I am sorry but I can not answer that, but Mr. So and so may be here later, and he can tell you" — that prospective customer is not going to come back to see Mr. So and so, nor return out of town. Of course, it is not necessary that Mr. So and so be assigned to an exhibit, but it is advantageous if at all times there is some one who can answer a customer's question. And, incidentally, not only be prepared to answer questions, but to rebutter inquiries.

Another area lacking at the Detroit Show, and one which was also lacking at the 1926 show, was uncorrected displays. Out of the 193 planes exhibited there were



Above: The Great Lakes trading plane mounted in flying position upon a revolving platform.



Right: The Ford & Whitney Aircraft Co. booth displaying a ground and a three-engine "Waco" and a ground and a three-engine "Beech".

not even had a *disarm* strapped fuselage and wings. As evidenced by the crowds that did inspect the few air-covered wings and fuselages, it is our belief that that is an important factor in aircraft shows that is being overlooked by the majority of exhibitors. It not only gives the prospective dealer and distributor opportunity to inspect the good workmanship of a product, but it also gives the layman and prospective retail customer a chance to note the sturdy and well-built construction of the modern airplane.

That fact is also true regarding aircraft engines. Believe it or not, there are many members of the aircraft

During other times they would not be admitted unless they were with an adult person. On the two "open" afternoons the exhibitors could watch their exhibits for "birds," and at all times be ready and willing to answer those "foolish" questions, and thereby give a boost to some unobtainable clasp.

NOT LEAST among the place exhibits that attracted the attention of the visiting public and members of the trade were the Fokker, Stout and Kreutzler displays which were grouped together in one section of the Hall. Here the layman was able to get a clearest view of the most modern means of commercial transport, and thousands of noses were pressed against the glass windows to get a better view of the inside of the passenger quarters.

A cross-section of the cabin of the forthcoming new Fokker 32 passenger transport attracted all kinds of attention as it was arranged that the visitor could walk right through and actually touch things. In addition to sleeping berths, etc., there were chairs so arranged about a little wall table that the air travelers could pass the time of day with a few rubbers of bridge. A remark was made by one visitor that actual bridge had its advantages in that when the leader saw that he was unable to make his bid he could signal the pilot who would immediately go into a steep vertical bank, spill the cards on the floor and thus make necessary a new deal.

One rather interesting fact regarding the Detroit Show was that there were more planes at the Ford Airport than there were inside Coney Island. A trip to the Ford Airport was an aeronautical education in itself, as there are listed almost every type of airplane imaginable, and most of them in the air at one time or another. Had there been some means whereby such viewing place was assigned to a certain part of the museum it would have helped materially in locating the various pilots and planes. And had there not been such an apparent vehicle-



A New Ford 8-1500 "Wasp" engine. Exhibit for transport airplane in flight.

industry to where the inside parts of a radial engine are in a certain means a complete mystery. At the Wright Aeronautical booth they and the public alike were able to view a cut-away Whitford in operation and see what it was all about. One or two other engine manufacturers also exhibited cut-away models and the crowds show that indicated their value as an aeronautical exhibit.

While the arrangement of aircraft and equipment exhibits was an improvement over the 1928 show, the individual exhibits did not show off in any great advantage. One of the reasons for that was the relatively small amount of space taken up by each exhibitor. Another was the noticeable absence of "working" models. There was, perhaps, an increase in the amount of literature over the 1928 show, but it is sincerely hoped that aircraft show exhibitors, particularly airplane exhibitors, will do more about reading material in the future.

One complaint that was heard quite frequently among the exhibitors was directed at the army of men who visited the show and attempted to walk off with everything that was not nailed down. To a certain extent the complaints are justified. A small boy at an aircraft show is, figuratively speaking, as big as a man, and he will try to get hold of everything he can, and perhaps justify the exhibitors with what may appear as foolish questions. On the other hand, it is that type of boy who will be buying planes when he grows up, or else become actively engaged in the industry itself. Whatever we can do for them now is, in a sense, building for as even better aircraft industry in the future.

To alleviate the "money bag" complaints that existed at the Detroit Show, it might prove wiser while to set aside two afternoons during which, boys under a certain age could be admitted accompanied by an adult.



The booth of H&H Industries, Inc. The H&H outboard pump "Barnes" engine on the left and the Wright 41 on the right.

side looking of the Ford Airport flying regulations it is quite possible that more place prospects would have been contacted in dissemination rights.

Although there were innumerable manufacturers and dealer meetings and other closed meetings held at various times during show week, there were several public meetings that proved to be of considerable interest to all present. What appeared to be the most important from the standpoint of subjects discussed and attendance was the flying school conference held on the morning of Friday, April 12.

For some time a committee headed by Phil Lee, has been studying the flying school situation in this country

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and cooperating with the Aeronautics Branch in an endeavor to straighten out school problems and place them upon a more sound and general basis. In order to do this it was decided to give Department ratings to all schools as being Approved or Unapproved schools. Therefore a set of regulations governing the operation of approved schools was drawn up, and it was that set of regulations which was presented to the members of the industry at the Friday morning meeting, in order that they might offer suggestions or criticism as they saw fit. The regulations are now being printed in manuscript

Below: A section of the aircraft display board in exhibit of Cordell Kirtley before 1928.

Right: The booth of Northwestern Airways, Inc. For the outboard pump on the table.



form and will be distributed throughout the industry within the next ten days.

One of the most important requirements stated that a school applying for Department approval be classified as Private Pilot, Limited Commercial Pilot or a Transport Pilot school. A second was the equipment to be used by each type of school. In this respect it was stated that there should be at least one plane for every 15 enrolled students and that approved schools could only operate from a field where there were 100 acres for every 30 planes flying off that field.

Another regulation was that every instructor must be a transport pilot at least and that the flying instructor's rating would go along with his transport license. In other words, he received at the same time and subject to the same classification reasons as a transport license. No instructor of an approved school is to be allowed to instruct for more than a period of six hours per day. There was considerable discussion regarding that point, so it was believed by query that a maximum of six hours was too much. Mention was made of the fact that three hours was the maximum in the Air Corps, and that the rating as it stood would make it possible for a new flying school owner to leave his instructors to spend six hours

per day in the air with a student. On the other hand, it was pointed out that a school owner would not take the risk of losing equipment by leaving a tired out instructor at work. As the result of the discussion it is possible that the maximum of six hours may be reduced somewhat in the final regulations.

The minimum times to be allowed for the completion of the various ground and flying courses were stated as follows:

Private license: Twenty-five hours of ground work, 10 to dual and eight hours solo. AE to be completed in

three months if the school is to recommend the student for Department examination.

Limited Commercial: Fifteen to twenty hours dual, and 10 hr of the required solo time must be on a cabin plane and other than the training plane. Fifty hours must be spent on ground work and the entire course completed in six months.

Transport: Thirty-five to fifty hours dual, 10 hours of the required solo time must be spent on other than the training plane, and 10 hours of the solo time must be spent on at least a four place cabin plane. A total of at least 100 hr ground work is required and the entire course must be completed in 18 months.

In order to hold a Department approved school using the school must have 60¢ per cent of its recommended commercial plane because standard quality. Department inspection are to begin upon approved schools on May 15. Information regarding other matters held during the Show will be found in the news section of this issue.

In summing up the Detroit Show and "convention week" it might be stated that with the facilities provided the Show officials did a rather good job. Had they provided a rest, a back room and, by all means, some way of finding the various exhibitors, etc., that would have helped. And had the exhibitors paid more attention to the visitors, and undoubtedly prospective customers, that stayed by their booths, their financial return from the Show would probably have been considerably greater than it really was. Just what the financial value of the Show was to the various exhibitors is a matter for considerable speculation, in view of the fact that many of the sales announced during the Show were arranged for before the doors of Coney Island Hall were open on Saturday evening, April 6.



# Export Sales Budgets

By WESLEY FOWLER

CONSIDERABLE thought is being given by United States aircraft manufacturers to the amount of money which can be spent for selling purposes. Production schedules have been drawn up and from all indications the total number of planes which will be produced in the United States during the current year will be approximately 8,000. It is believed that during 1939 a production figure of 10,000 will be reached (These are conservative estimates, there was talk of a production of 8,000 for 1938 after last year and the talk now is of 10,000 for 1939.)

Few manufacturers have considered what the foreign market will absorb in the next few years and the aircraft industry, therefore, is at a loss to know how far it can go in spending its hard-earned cash for foreign advertising, demonstrations abroad and for general pioneering to get its share of foreign business. It is reasonable to believe that 30 per cent of the United States aircraft production will be exported, which would indicate that of the 8,000 airplanes produced during 1939, 800 should find their way overseas and to the bordering countries. Our factories have not as absorbed with problems of production, and some with meeting the demand within this country, thus little or no energy has been expended toward securing foreign business. Of the 4959 military and commercial planes produced during 1938 only 170 planes were exported—about four per cent—and most of these were sold more by accident than design. During the latter part of last year a few of the larger manufacturers became interested in the possibilities of foreign business.

They, however, have been working largely in the dark because there has been a lack of adequate data and particularly no precedent upon which to base their activities. A few occasional foreign orders have come to them as the result of efforts of American Consuls and the Bureaus of Foreign and Domestic Commerce.

AN ATTEMPT will be made here to analyze the business as it existed from abroad in the past four years and to budget what may be expected in the way of business during the next two years from the various countries not actively reinforcing the importation of American aircraft in this last connection, there are certain countries now requesting the importation of our aircraft which are expected to lift the lid within a short time. These nations, therefore, will be included among those for which market opportunities are forecast.

It should be stated here, that it is impossible to budget foreign business on the same way that production and domestic sales quotas are calculated, namely, on the basis of the quota clauses contained in distribution and distribution contracts. The data foreign distributors for aircraft have been found to be reluctant about signing a contract which embodies a quota clause, and it is believed that

# for AIRPLANE MANUFACTURERS

FIG. 1—U. S. AIRPLANE EXPORTS  
Quotations of 1938, 1939 and 1940

Country of Destination	No.	Value in Dollars	No.	Value in Dollars	No.	Value in Dollars
Canada	12	200,184	10.7	26	211,247	11.3
Panama	47	230,185	6.1	10	20,000	11.4
Argentina	25	170,000	6.0	1	10,000	1.1
Mexico	1	144,000	6.1	9	14,000	14.5
Costa Rica	27	144,000	6.1	9	14,000	14.5
France	4	100,000	10.2	4	10,000	11.2
Australia	2	24,000	4.7	7	50,000	11.3
United Kingdom	15	100,000	4.6	4	20,000	11.4
China	6	210,000	3.9	1	10,000	11.4
Germany	1	10,000	3.9	1	10,000	11.4
Spain	1	10,000	3.9	1	10,000	11.4
Belgium	4	10,000	3.9	1	10,000	11.4
Japan	1	10,000	3.9	1	10,000	11.4
Switzerland	1	10,000	3.9	1	10,000	11.4
Netherlands	1	10,000	3.9	1	10,000	11.4
Sweden	1	10,000	3.9	1	10,000	11.4
Italy	1	10,000	3.9	1	10,000	11.4
Portugal	1	10,000	3.9	1	10,000	11.4
Poland	1	10,000	3.9	1	10,000	11.4
Czechoslovakia	1	10,000	3.9	1	10,000	11.4
Yugoslavia	1	10,000	3.9	1	10,000	11.4
Romania	1	10,000	3.9	1	10,000	11.4
Greece	1	10,000	3.9	1	10,000	11.4
Other countries	1	10,000	3.9	1	10,000	11.4
<b>Total</b>	<b>305</b>	<b>\$1,000,000</b>	<b>3.9</b>	<b>100</b>	<b>\$1,000,000</b>	<b>11.4</b>

\*Minimum to qualify as foreign export on production line of this figure.

FIG. 2—U. S. AIRPLANE EXPORTS

Country of Destination	No.	Value in Dollars	No.	Value in Dollars	No.	Value in Dollars
Canada	12	200,184	10.7	26	211,247	11.3
Panama	47	230,185	6.1	10	20,000	11.4
Argentina	25	170,000	6.0	1	10,000	1.1
Mexico	1	144,000	6.1	9	14,000	14.5
Costa Rica	27	144,000	6.1	9	14,000	14.5
France	4	100,000	10.2	4	10,000	11.2
Australia	2	24,000	4.7	7	50,000	11.3
United Kingdom	15	100,000	4.6	4	20,000	11.4
China	6	210,000	3.9	1	10,000	11.4
Germany	1	10,000	3.9	1	10,000	11.4
Spain	1	10,000	3.9	1	10,000	11.4
Belgium	4	10,000	3.9	1	10,000	11.4
Japan	1	10,000	3.9	1	10,000	11.4
Switzerland	1	10,000	3.9	1	10,000	11.4
Netherlands	1	10,000	3.9	1	10,000	11.4
Sweden	1	10,000	3.9	1	10,000	11.4
Italy	1	10,000	3.9	1	10,000	11.4
Portugal	1	10,000	3.9	1	10,000	11.4
Poland	1	10,000	3.9	1	10,000	11.4
Czechoslovakia	1	10,000	3.9	1	10,000	11.4
Yugoslavia	1	10,000	3.9	1	10,000	11.4
Romania	1	10,000	3.9	1	10,000	11.4
Greece	1	10,000	3.9	1	10,000	11.4
Other countries	1	10,000	3.9	1	10,000	11.4
<b>Total</b>	<b>305</b>	<b>\$1,000,000</b>	<b>3.9</b>	<b>100</b>	<b>\$1,000,000</b>	<b>11.4</b>

\*Minimum to qualify as foreign export on production line of this figure.

ered by most manufacturers, even of products other than those aerodynamic, as part of the domestic market, should continue to be an excellent field.

In the estimates for 1939 and 1940 (Fig. 2) it was considered that several important manufacturers had established plans there which will mean that most shipments of airplanes will not be shown as such in official statistics, but will be included under the classification of "airplane parts." The 15 flying clubs in Canada are now being equipped with British planes. It is reasonable to suppose that when some of our light and inexpensive training planes, with new production engines, have the "Bugs" taken out of them, that they will be sold for club use in Canada. For this reason the unit value estimate for 1940 is shown to decline in the case of Canada.

Pure is a country which although undergoing difficult times from an economic standpoint could use air transportation as a means toward economic recovery. The present aviation operations in the country and the advent of the international airline from the United States indicate that the sales forecast for this country is conservative. Because of the difficult flying conditions in Peru sales of low priced planes should be in the minority.

THE NEXT country listed did not present a very large market during 1938, but 1939 starts off well with the sale of one American plane to the Argentine, valued at \$50,000. The few sales made there last year point to an opportunity for considerable expansion. Brazil is a wealthy state which has signified its interest in aviation and is willing to be sold further on the idea. These have been no American commercial airplanes demonstrated in this market.

Mexico is another country suffering from economic instability, and there is every reason to believe that a few more air services would help it out of its present trouble. There are several lines now in operation, all of which are American equipment. It is expected that early this year the Government will have canceled the purchase of some American military aircraft. Some of the resumed exports will undoubtedly be to the American operating company, but they nevertheless will be sales of aircraft far in excess of those which will contribute to the prosperity of the country and to the culture of our airplane factories.

There is now some important business pending in Brazil for leasing planes and aircraft for transport purposes. It is thought that the first manufacturer to demonstrate plans of this type in that country will find the effort worthwhile, if not from the immediate orders from the standpoint of having made a start in the market. It will be noted that the estimated average valuations for most countries show a decrease for 1940 under those for 1939. Brazil, as well as some of the other countries, may be considered something of a price market. Furthermore, by 1940 it is thought that in-

Aerometric Section of the Bureau of Foreign and Domestic Commerce, Department of Commerce, on market conditions both favorable and inhibiting against the importation of American airplanes. The estimated valuations of certain countries may seem unreasonably optimistic. It is believed, however, that a marked amount in each country for the approximate number of planes listed for each. Details during the next few years may not bear out these figures but by adequate and appropriate sales effort an individual manufacturer should sell as much as he can to each market. This sale should be on the basis of the percentages of a specific factory's production to total production, i. e., if 800 airplanes are produced by one factory, 80 should be exported, 40 airplanes are budgeted to total sales to Brazil during 1939, therefore, this manufacturer should sell at least 40 to that country.

By grouping the countries geographically (Fig. 1), we find that Latin America, including Mexico, the West Indies and Central and South America, absorbed 70 per cent of the exports (exclusive of those to Canada); that the Far East including Australia absorbed 20 per cent and Europe 10 per cent. Canada, which is consid-

ered to be a market before market absorption in foreign countries can be estimated upon the basis used in this country.

The following analysis is strictly of a tentative nature, with stress on the tentative, because of the many uncertainties entering into foreign marketing such as the varying financial conditions of the respective countries, the numerous restrictions which may be placed against our aircraft, etc. It is believed that the following survey will have a practical application toward solving the export problems of the individual manufacturer. For example, if one plant schedules a production of 800 aircraft for 1939 and the total airplane exports from all U. S. factories during that year are 800, or ten per cent of total estimated production, that manufacturer (who should export 80 airplanes if he is to get his share of foreign business for the year) would like to know on what foreign markets to concentrate, where to send representatives and demonstrators and where to place his foreign advertising. The following sales forecast by countries is based upon exports of airplanes from 1939 to 1938, inclusive, and data obtained from the files of the

Under this heavy pressure of activity, the Department has never caught up. The various divisions and sections are still trying to find "their own" to eliminate delay and get things done for the air—and to keep them there. This has been the watchword of the Aeronautics Branch since its beginning. It has never been a red-tag office, in the accepted sense.

**B**UT THE WORKERS of planes, pilots, mechanics and students have been and continue to be one big jumpy crowd. Naturally, there have been regrettable instances of lost business in the industry, embarrassing to manufacturers, operators, pilots, schools, and students, simply because the Department has not had enough personnel—both field and office—to keep abreast of the work coming. Nor could it "bite" on its defenses, and broadcast this condition to the industry. That is, no one on the "inside" could do so, and no one on the outside appeared the unfortunate situation.

In the past two years some chronic "irritants" and some new with legitimate complaints have irritated the Department lately, especially the Regulations Division, with its licensing, inspection, and engineering sections.

Those who complained very nearly had a chance to be free from the "intolerable burden"—and the danger is not permanently past.

Here is what happened:

The Budget Bureau, accustomed to large demands and quiescent countries by Government departments, and quite accustomed to cutting these demands and turning a deaf ear—failed to see any economy for a certain line in the appropriation bill for the Aeronautics Branch "Traveling expenses for inspectors." This item was to cover the expense of inspectors in making their regular rounds through their districts. The Department had found that regular itineraries, published throughout the district, permitted more pilots and owners to be at specified places on certain dates and be examined, or to have their planes inspected. Only by these regular trips, complete with a quarter and bags, again at once, could licensing and inspection remain even in "one step ahead."

**B**UT, said the Budget Bureau—in effect, "let the inspectors say 'yes.' There's no need of their coming around all the time. Let the pilots come to them at one established base."

That may not at first glance seem serious. But it reduces to this:

The Department could not reasonably compel an applicant to leave his place of business at a time it might specify his plane a long distance, and wait, perhaps several days "in line," for an examination. If it did it would tie up hundreds of planes, hold pilots on the ground, cause a hopeless jam at the inspector's post, and anger the industry. It would have to "inspect" applicants to prevent them from a certain date, or "look a time at a convenient," meanwhile giving letters of deficiency as was first done in registration work. These letters of deficiency would run on and on, with renewals, until the applicant found it "convenient" to come for his examination. Any economic excuse would have to be accepted. Hardly any consent on this is needed. Even the best intended pilots would wish their own pleasure to "drop in and see the inspector." Those who had any doubt about their ability would be strangely busy at some other part of the district. Months would pass before they

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appeared, if at all. And even strict letters, ordering them to appear, would not help much, unless an inspector had authority to go out and look them up.

The same applies to airplanes. When and if the owners cared to fly their planes to the inspector they would be inspected, not otherwise. All the time, letters of authority would flourish abundantly, the standard status of the industry would hang long and merely at the Department of Commerce.

Until capital withdrew.

For whatever it would, unless the industry got busy at once and set up its own standards, or carried out the Department's system. That is hard to expect of any industry, especially one growing so rapidly as aviation. And it would take time. During the period of change there would be no one.

The standard components would be the first to enter their doors to aviation. Standards would be too uncertain for writing big policies. They would not take the word of operators, unassured by other opinion, official in nature. Rates would soar, high as those of 1918 and 1919, perhaps higher. Private companies would want all airplanes telling from their books, until the cost was too. Operators would again have trouble in getting good steady pilots—and making them stay that way. And in the piloting game there would be one grand few-far-all. A student could run wild, taking passengers where he pleased, as long as he stayed a short distance from the inspector.

**T**HAT IS WHY that this opinion would not be too far-fetched, I asked the question first mentioned in this article of Maj. Clarence M. Young, director of Aeronautics, Department of Commerce.

"If regulations were to come for six months, we'd be worse off than before we started," he told me. "It would be chaos and the worst kind. And it would take twice as long to get even with what we are now. And we aren't caught up yet."

"This is no cry of 'Wail.' Nor is it any more criticism of the Budget Bureau, for naturally those not close to the industry cannot begin to appreciate what a disaster this step would be. Yet the outlook is none too bright, for this is a period of utmost public confidence in flying. The business is popular. And if the picture is not clear only to those who can so easily multiply negatives, it may be less clear tomorrow."

The Department of Commerce can do no more than it has done. Desperately, with strong loyalty to the industry, it has tried to point out what would happen under such circumstances. But, the Budget Bureau is not to strong place, and probably it is unconvincing in suggesting personal desire to build up departments. The Department should have the heavy support and co-operation of the industry in this matter. The industry has been too busy in the "paid role" to learn these problems. It has been glad to answer what help it could from the Department. When it has been held up for any reason it has complained—naturally. But a little study of the situation, and understanding of support in the request for more field and other personnel would result in discussion of the trouble—and result in direct benefit to those governed by the Air Commerce Act.

This is the first of a series of air articles prepared by Mr. Krynke and dealing with the Aeronautics Branch, Department of Commerce. The second article will appear in an early issue.—Ed.

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# The TREND in AIRPLANE Body Design

By H. C. WENDT

Body Engineer, Fairchild Airplane Works Corp.

**A**NY well-informed automobile buyer will tell you that the purchasing of an automobile is usually a fairly easy job. For this reason, he will also tell you, 90 per cent of automobile sales are made through women.

He will tell you that most women have little regard for mechanical excellence, performance or speed, but base their judgments mainly upon appearance and comfort.

What is under the hood and in the chassis is not well understood by them and is a matter of minor concern. Being open-minded to the fact that all makes of cars seem to carry their passengers from point to point without noticeable effort, they are convinced to let the matter remain there and take performance and mechanical perfection for granted. Let the exterior present a study in graceful, well-proportioned design, however, let the point combination be particularly pleasing and the material a combination of luxurious upholstery and rich appointments, and the decision is made and the car is good as sold.

The influence this attitude has had upon the automobile industry cannot be estimated. We do know that if automobiles had remained in the form in which they were 10 or 15 years ago, we should never have seen the automobile industry as we know it today, regardless of the degree of mechanical perfection attained. We also know that if the value had not descended its ugly path covered with dust and vermin, if the phone-rickshaw had not conquered its mechanics and clumsy body and if both had not become articles of furniture which inter-



In interior view of the cockpit of a Fairchild 107 showing the leather seating, built-in automobile features.

mediate in the most fashionably furnished home, these would not have been the complete public acceptance they enjoy today. This trend toward refinement of the product is almost universal. Even such everyday articles as the electric toaster, vacuum cleaner, typewriter, kitchen mixer and what-not have come under this influence.

**N**OW that women have started the reform, the men are wholeheartedly helping the thing along and will eventually take the entire credit to themselves. Without blurring the line to prove it, we can safely state that the mechanical automobile from primarily toward utility while the latest trend takes pleasure and comfort and has more trust in a product that is a bit the part. What can we do?

First, we must realize that the automobile industry has been here for 25 years. Certainly it will have no influence upon the types of places which carry our mail, express and freight. Neither will it affect those places which are used for combat purposes by the Army and Navy, nor highly specialized types used for endurance flights, exploring, troop dumping, photography, sniping and the like. But we cannot be content with any such limited usage as this. We cannot be content with anything short of complete public acceptance. This means that facilities and individuals must make the same use of airplanes as they now make of automobiles, buses, railways, steamers and other means of transportation, and do it just as enthusiastically. We know that day is coming; inevitably it must. But how can we accelerate its arrival? Simply by making

confidence in our product into the public, now more receptive than ever before.

I know of one transport company which lost five passengers in a 400 mile trip, just because one member of the family had lost confidence in the equipment. One look at the plane, this company said, just a week before the family vacation began, stifled the question. The entire family started its vacation with a true ride.

Last summer I spent many of my leisure hours at various flying fields and at times am privileged to have a close-up view of the public's attitude on flying.

I remember one young lady who refused her certain offer of a trip with the remark: "No, I'm afraid. That plane looks like a starved horse with its ribs sticking out."

Another declared with "The black thing looks like a big coffin." These are not isolated cases. I could quote many others from my own and others' experience. Some say that women are fairly that nervous however, it is not an answer nor a solution.

The reason that eye appeal is such a great factor with this sex is not a great mystery. I believe that centuries of training along non-aeronautical lines has forced this trait upon women. Man is entirely confined to women. Anyone who does not understand the manner of operation of a machine must necessarily let his judgment be based upon the evidence of his senses. A very small proportion of the present day public understands the theory of flight or the operation of aircraft themselves. Eye appeal cannot help but be of the greatest influence in establishing confidence.

The airplane, naturally has very little exterior resemblance to any other means of transportation. Instead of fostering that dissimilarity, we should do all in our power to lessen the difference in appearance. As the automobile is far preferable to any other means of travel, it should serve as our starting point in exterior design. As soon as our planes strike a few notes in common with those of the automobile they will also strike familiar chords in the minds of the public. The top becomes "motor compartment" and "air-minded" will be helped by any stages of every surrounding is made transparent of the main fuselage design.

Possibly the extent step in this direction that can be made is in the paint scheme. In no other branch of transportation, it fact is no other product offered to the public, have we seen such wild and impossible combinations of color. We have on the average airplane today. Visible, of course, is a factor but I do not believe that the familiar yellows and oranges which we have used on the fuselage and wing are necessary. Why not use a color which will be visible enough

and still neutral to any other color which might be used to decorate the plane? I believe this requirement is easily met with a fairly deep shade of cream. This color is highly visible and yet blends nicely with red, blue, and practically every shade of green and blue. Having painted the wings and possibly the tail surfaces in this shade of cream, let us decorate the fuselage in two contrasting shades of blue, or green, or in a deep maroon and a lighter red, taking care always to keep the darker color above the lighter color, as this reduces the apparent height of any object and accentuates its length. Let our colors be bright if they must, but let us not violate conventions and good taste by piling on layers of blue, orange and scarlet with absolute disregard to every law of color harmony.

Consider doors and windows for a moment. Doors certainly must be wide and high enough to permit of easy entrance and exit. That there must be given no adherence to the fuselage structure, in order that this requirement may be met and that no rules cut off the corners of the doors, thus producing unnecessary shape and inconvenience to passengers.

Before any tubular or other structure is definitely decided upon, every effort must be made to arrange the structure in such a manner as to eliminate unnecessarily wide gaps between windows and to guard against any of the construction's showing in the window opening.

Other structure is definitely decided upon, every effort must be made to arrange the structure in such a manner as to eliminate unnecessarily wide gaps between windows and to guard against any of the construction's showing in the window opening.

**F**ew reasons have any idea of the loss which may be safely imposed upon a tube. When a tube is exposed, their thoughts run to the apparent dimensions of the structure and their confidence in it is destroyed.

In the design of an automobile, window form and arrangement is considered one of the most important details. The same must be made true of an airplane if we are to reach a similar state of perfection. Windows should be long and narrow in order to accentuate the apparent length of the cabin and reduce the apparent overall height of the fuselage. Frank shapes which do not blend in with the rest of the design must be eliminated. Each window should be framed with beautifully finished sheet metal so as to avoid the metal. It is sufficient to make this a flat piece of metal resting flush against the window. The windows should be set in at least a short distance and the metal flange in it to meet it, in order to give the appearance of solidity and to eliminate the "finny" appearance which a real edge of sheet metal always gives. A thin metal molding just below the windows and raising the full length of the fuselage, outside, this molding actually existing and not merely painted on, will again add to the apparent

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steady length and will form a logical landing point for the two shades of color used on the fuselage.

Another word about doors. Psychologically a door half-inch out of its center is very bad. Frequently people vainly consider the door as the only thing between them and a drop of some (loosened) fat. A light but thick and bulky looking door will give an added feeling of security far out of proportion to its cost and slight extra weight.

Having attracted and harmonized the outside appearance of the plane, and thereby having provided the public with an easy mental stepping-stone to a comparatively new type of transportation, we turn our attention to the interior. Here, indeed, we have an opportunity to do wonders.

To begin with, we find that the average plane, particularly the cabin plane, is entirely too narrow for two persons to sit in side-by-side in comfort. Before the public will generally accept the airplane as a means of common transportation, fuselages must be widened so that the seating comfort is even greater than it is in the automobile. Nothing is more uncomfortable than a long trip in close quarters. Passengers get up with such a thump on a short bus taken just for the thrill, but the airplane industry cannot exist on the basis of selling but one ride to a person. We must sell the public on the idea of using the airplane to go anywhere, regardless of distance, and to do so we must provide a degree of comfort which compares that of any other means of



The interior layout and the outside of the new Fairchild "11" cabin monoplane.

transportation. The upholstery must be not only comfortable but luxurious. Passengers must be able to lounge at ease, move about when necessary or disengage, read, converse, do off, or watch the countryside slip by. Everything in the cabin must convey a feeling of ease and comfort.

The floor must be covered with a rich carpet, the cushions with well-stuffed headrests, leather or any of the latter types of materials found in the finer motor cars. Door handles, window handles and other hardware must be rich in appearance; wadding mats, arm supports and the other relaxing details of the finer cars must be included. All tables and other structure must be set up, the windows finished with walnut or mahogany moldings, the doors weather striped in elegant designs, and the windows installed in such manner as to be easily adjustable for

ventilation and yet absolutely weather-tight in cold or storm. Other relaxing details such as brochures, standard, sewing cases and playing lists should be added to create the impression of safety and insure attention to detail.

A great amount of study and work must be done to eliminate the various and numerous sources of noise in the interiors of cabin planes. Engines must be shielded so that conversation can be carried on without undue the greatest discomfort experienced in air travel is the terrible din which assaults the ears at all times. The public will not tolerate this very long, and unless every effort is bent towards eliminating this unpleasantness we will find it very difficult to interest the passenger in a second ride. The use of silencers on engine exhausts, and the greatest sound absorbing and sound reflecting materials around the motor cabin have proved effective, under circumstances can be the use of such materials in lateral quantities be dispensed with in passenger planes.

**O**UR CASE I have heard many arguments against any procedure so I have outlined above. One argument is that it adds weight added weight makes for decreased performance, which again means fewer rather than more sales of such a revised plane because in the first analysis the pilot must be able to handle the plane with judgment against air performance. It is true that it is to extend the report of a pilot has a heavy bearing in the purchase of a plane, but I rate consideration of every pilot in the country by seeing that it should not.

No doubt a good pilot is a better person to pass upon the mechanical condition of aircraft than, let us say, the executive of a transport company is the market for planes to be used in passenger service over an air route. But the features which will it in the pilot are not the same features which will it in the public. What if a good wing makes does decrease the speed of the plane a few miles an hour compared to one with a narrow, cramped cabin? Will such a consideration enhance a passenger plane to look like a four or five-hour journey? News in the world?

I realize that a certain proportion of the society-savvy, thick-lipped public will fly in a plane regardless of its appearance, refinement and comfort. While this percentage is poor, that is, we cannot count upon it to be without falling short of the complete public acceptance which we seek. It is certain that the sale and use, constructive type of effort will not purchase an airplane for personal use unless it compares favorably in appearance with an automobile of equivalent price. Mechanical excellence, of course, being taken for granted. Nor will these good folk use an airplane in preference to other means of transportation unless we give them the more degree of comfort plus greater speed, then they can get in any other sort of conveyance.

Some men in the airplane industry agree that the necessity for refinement and beautification is true in principle, but say that the time is not yet ripe for it, and such treatment must be reserved for the future.

I cannot agree with this statement. The airplane has proved its utility. Fashion fashions have put the public in a receptive mood towards aviation and public interest is rising. Let us make their improvement now, before the public comes to the conclusion that the old, comfortable, slower means of transport are preferable, after all.



Exterior view of the Fairchild "11" cabin monoplane. This photograph was taken from the pilot's and looking back.



## Fairchild States Expansion Program

Address: Detroit  
Gathering of Agents

DETROIT (Wichita)—An expansion program of wide scope was announced by Sherman M. Fairchild, president of Fairchild Aircraft Corporation, at a dinner given by the corporation to its distributors and dealers at the Sheraton-Chester Hotel, April 14.

A. A. Woodruff, sales manager, outlined the sales and service policy adopted by the corporation, emphasizing adaptability, efficiency and the advantages of Fairchild models 21, 41 and 71 and the Republic-Brewster Challenger Indians which have been added to the Fairchild line.

Information conveyed to the dealers indicated that the Fairchild Corporation, by reason of the varied manufacturing and service activities of its subsidiary companies and by increased collaboration with the Aviation Corporation, was occupying a position of great strength in the industry.

### Increasing Flying Hours

The Aviation Corporation, which may control 50 per cent of the Fairchild stock, now holds a large interest in Universal Aircraft and Colonial Industries.

Fairchild Aviation has now acquired the Knudsen-Bessner Aircraft Company and is interested in the Piershagen Aircraft Corporation, which will produce an aircraft plane.

It was announced furthermore that Fairchild was conducting extensive experiments with flying boats and amphibious planes, and will in the near future, announce a significant development along that line.

Announcement was also made of the world's largest commercial wood frame, fabric-covered biplane, the Fairchild 41, in charge of the Army Air Corps Engineering Division at Dayton, Ohio, and is to be installed at the Fairchild plant.

### To Produce 100 Planes

Last year the Fairchild corporation and subsidiaries had a gross business of \$2,600,000. This year, including production of Republic-Brewster planes, Fairchild expects to do a gross business of \$4,000,000.

The plane production schedule, as announced by Mr. Woodruff, will include Fairchild planes, which included 300 20's, 300 41's, 100 71's, and 300 Republic-Brewster planes. As an indication of the greater demand, Mr. Woodruff stated that these five orders, totaling 1,000 planes, had been made in the factory since the beginning of the year.

Cooperation with dealers, Mr. Woodruff said, called for an appropriation of \$25,000 for technical advice and

travel, and an auto-plane house organization is to be distributed monthly. A delivery scheduling system, which serves representatives, a substantial contribution to dealers on aerial supply contracts, and a check in addition to rental of dealer's planes for the aerial service work, was also announced.

In addition, Fairchild adopted a modern training school in Farmingdale and will distribute to dealers all possible training material.

### Model 71 Product

Delivered within 60 days were promised for the new model 71, designed as a four-place dual control with 225 hp. Whirlwind, or five place dual control with 300 hp. Whirlwind, with an unimpaired performance equal to the leader in the market. Mr. Woodruff stated that Army tests recently completed, showed the Model 71 as an outstanding performer in its class.

In discussing available facilities for the manufacturers of the 40 hp. Gnat engine, it was said that Fairchild has an engine plant open to any in the industry from which some very interesting technical developments may be expected. The Gnat engine was reported to be 200 hp. of dynamometer and flight testing of the Gnat engine, during which time only two minor failures occurred.

The assembled dealers were given a very frank statement regarding actual costs manufacturing methods employed by the corporation recently, namely before the California engine is come up to expectations and innovations in design of models 41 and 71 are primarily to stability to obtain the greatest performance in design.

### Engine Inspection Technique

Throughout the experience gained and the modern engine production tools and complete engine testing apparatus installed at Farmingdale, which are now available to producers of the Gnat engine and for other important engine developments are proving of great value to the general engineering program.

The Canadian engine tests resulted in bringing into the Fairchild group the Continental engine Corporation, because Mr. Woodruff stated, the plane produced by this company proved to be capable of withstanding the excessive vibration of the Canadian engine.

### Guts Aircraft Plant Contract

WICHITA (Wichita)—The W. S. Hewson Construction Company has been awarded the contract for construction of six new brick and steel factory units at the Geneva Aircraft Company building, project, hereinafter \$1,250,000. Plans call for a building by the middle of June. It will contain 80,000 sq. ft. of floor space. The Geneva company is a firm in business twenty and a half miles from Chicago, and has a four and a half place Curtiss Wright airplane recently after moving into the new building. The payroll will be increased to approximately \$100,000.

## Aviation Corporation To Absorb Colonial Line

NEW YORK (Wichita)—Colonial Airways Service, which includes Colonial Western, Canadian Colonial, and Colonial Air Transport, is to be absorbed by The Aviation Corporation and will report here. A contract between the two concerns is said to have been completed and completion of the merger hinges only on the acquisition of some \$100,000 out of the Colonial stock.

The announcement was made on the eve of the completion of the Colonial's first daily line New York-Boston service arriving today using four Fairchild planes. The Aviation Corporation controls about twenty planes over 900 cubic feet of average carrying load and passengers. The line connects Boston and New York, Cleveland and Montreal and Albany and Cleveland by way of Buffalo.

May Gen. John P. O'Connor, head of Colonial, will continue to direct the system to vice-president at The Aviation Corporation and a member of the board of directors and executive committee. The Colonial's treasury is capitalized at \$100,000. It has been operating between Boston and New York since July 1.

At the same time it was revealed that the Aviation Corporation is planning to use the former Colonial's Embury-Baldwin Company (Chicago), which operates a mail and passenger line between Chicago and Cincinnati by way of Indianapolis. This company is at the Chicago with Universal Aviation system and at Cincinnati with Continental Airways. The Aviation Corporation is said to have been formed under a Delaware charter, which is subject to the laws of that state. The Aviation Corporation will have control of this company with Air Investments, Inc., of this city.

### Light Lear Plane Tested

WICHITA (Wichita)—Powered with a Continental engine, the new Light Lear plane tested 32 hp. the new Lear large engine monoplane which it is to manufacture by the Lear Aircraft Company of Wichita. It is reported to have passed its tests with flying colors. The small winged craft, weighing 675 lb., with engine, developed a speed of 80 m.p.h. and landed at the low speed of 30 m.p.h. Charles E. Lear, a student of the University at Wichita, designed the plane.

### Fordian Staffed

WASHINGTON (Wichita)—Baldwin-Geisler, Inc., of Toledo, is said to be the chief of the Air Corps and as one of the pioneer aviators of the Army, has been assigned to duty in the military division. O. O. Smith of the Marine Division, effective June 30, and Maj. Gen. William T. Collins, also assistant to the Chief of the Air Corps, who is now chief of the Marine Division at Wright Field, has been assigned to report to Washington by June 30.

## Nine More Planes One Engine Approved

WASHINGTON (Wichita)—An announcement is made here by the Aeronautical Branch that nine more plane models have been granted approved type certificates, bringing the total certificates awarded to 127. Another engine, the Continental, has received approval, the power plant being the Wright R-540, five cylinder radial air-cooled engine. This plant, which develops 145 hp. at 2,000 r.p.m., was approved April 23.

The planes are as follows: No. 125—Biplane C-130, 6 place radial engine and monoplane, Wright R-540-A, developing 240 hp. No. 136—Trend Air R-6000, 4 place cabin land monoplane, Wright R-540-A, No. 137—Trend Model 2-2, 4 place open land biplane, Kinner R-4, No. 137—Trend A-1-A, 14 place cabin land monoplane, Wright R-540-A, No. 138—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 139—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 140—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 141—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 142—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 143—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 144—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 145—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 146—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 147—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 148—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 149—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 150—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 151—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 152—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 153—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 154—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 155—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 156—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 157—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 158—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 159—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 160—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 161—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 162—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 163—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 164—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 165—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 166—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 167—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 168—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 169—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 170—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 171—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 172—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 173—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 174—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 175—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 176—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 177—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 178—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 179—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 180—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 181—Beech R-500, 6 place open land 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244—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 245—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 246—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 247—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 248—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 249—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 250—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 251—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 252—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 253—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 254—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 255—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 256—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 257—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 258—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 259—Beech R-500, 6 place open land 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322—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 323—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 324—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 325—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 326—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 327—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 328—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 329—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 330—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 331—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 332—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 333—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 334—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 335—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 336—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 337—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 338—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 339—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 340—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 341—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 342—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 343—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 344—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 345—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 346—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 347—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 348—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 349—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 350—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 351—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 352—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 353—Beech 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400—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 401—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 402—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 403—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 404—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 405—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 406—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 407—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 408—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 409—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 410—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 411—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 412—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 413—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 414—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 415—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 416—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 417—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 418—Beech R-500, 6 place open land biplane, Wright R-540-A, No. 419—Beech R-500, 6 place open land biplane, Wright R-540-A









## FOREIGN ACTIVITIES

Mexican Aviation Co.  
Has Long Experience

**NEW YORK (A. V.)**—The Mexican Aviation Company which now is carrying the air mail between Brownsville, Tex., and Mexico City, in trans-gulf flights, the direct successor of a postwar corporation which started in 1939 under the name of Mexicana, has been awarded a contract to fly mail and cargo between the United States and Mexico. The title which the company now bears was applied only last fall when British blood through the Aviation Corporation of the Americas was brought into the original company.

The Mexican Air Transport Company, as it was known first, was organized to ferry passengers and company mail, and miscellaneous goods between Mexico City and Tampico for oil and oil interests. Following the growth of aviation, the company in later years added Pan-American airlines and now has entered the air mail business over expanding lines, the climax coming in March 3 when the Pacific west coast service on the Brownsville-Mexico City route.

Shortly before the opening of this line, the company began operation of a strictly Mexican line which is even longer than that to which it is connected between Vera Cruz and Mexico City, with stops at Minatitlán, Villa Hermosa, Ciudad del Carmen and Cancun. This is called *Avianca*. This line is to be used by the Pan-American and will connect also with the Pan American service between Mexico City and the Central Zone of Central, San de Mexico, as soon as a landing field has been prepared there.

## Dutch Use English Airports

**THE HAGUE (REUTERS)**—The opening of this country's subsidence has been planned to the Dutch East Indies which was started off at intervals last year and as a result of extremely unfavorable conditions, are to be used in the future by the use of English airlines along the airway, permission for which has been granted recently. One of the general obstacles encountered by the Dutch airlines on the long flight was the absence of suitable landing fields.

## English Air Mail Increases

**LONDON (REUTERS)**—About 30,000 lb. of letter air mail were flown from England in 1958 representing an increase of 36 per cent over 1957. Flares on the Cato-Burns service arrived about 20,000 lb. as compared with 14,000 lb. in 1957.

## Foreign News Briefs

Seven legs of 31 miles each around the equatorial course for the September Trophy Race at Geneva this year have been submitted for the line page originally planned.

Mexican and American capitalists are reported organizing a \$3,000,000 company to operate an air mail and passenger service between Rio de Janeiro and San Francisco by way of Mexico City and various cities of Central America countries.

Colombia is studying an ecological mission to this country to study aviation here. The mission will visit European countries and American also, it is expected.

France has launched its second airplane carrier, a 10,000-ton vessel accommodating 20 airplanes and flying boats. The ship is 347 ft long, has five elevator cranes for hoisting the machines on board and four catapults for launching the planes. It carries a crew of 43 officers and 600 men. The ship will have a speed of 20 knots.

Passenger service between Colombia and the Canal Zone was scheduled to have started Wednesday, April 17, by Seaboard. The company has been carrying mail between the Canal Zone and South American ports.

Kagman, Ontario, Flying Club has been added to the clubs of flying clubs organized in Canada under Government policy.

A French contingent has an outbreak of the principal cities along the country's airways in being given considerable expense.

Van Leer Black, whose Cape Town-London flight in his Transglobe P-51 Mustang, was forced down at the French-Italian border April 3, still plans to carry out his round trip flight in Japan and expects to take off in his repaired plane the latter part of this month.

There is agitation in Buenos Aires, Argentina, for construction of a dirigible line there in anticipation of trans-Atlantic dirigible service between that city and Seattle, Spain.

Geoside Service from mail has been under test recently at Stag Lake, Arizona. London, in both medium and high power engines under suspension of the British Air Ministry. The test, scheduled from late September to early October, will be conducted before the airship is conclusively judged.

The Royal Dutch Air Line has started a special flight service between Hol-

Italy Holding  
Light Plane Test

**ROME (ITALY)**— Fiat, Macchi and Caproni light planes are reported to be expected to be the three that will take part in the final test which the government is conducting to decide on the official light plane which it will furnish to the school flying schools and flying clubs service, however. The three planes taking part in the early tests, the Macchi, the Pirella, the Pirella, the Curtiss, and the Macchi.

The test includes a 100-hr. flight, a series of three flights in which the fuel and oil consumption are tested, tests on landing and take-off performance and speed in horizontal flight and climbing. The final test, in which an approved Italian test plane is used, will be a flight of about 2,000 miles around Italy. The engine were being tested for reliability in the approved plane and the Fiat engine under test of 45 hp, the Pirella, four-cylinder version of 45 hp and a Macchi four-cylinder of 20 hp.

## German Air Budget Cut

**BERLIN (REUTERS)**—Aviation in this country is believed to have received a severe blow with the cutting of the budget assigned to the aviation ministry. Last year's budget of 10,000,000 Marks has been cut in half, threatening the aviation ministry with a curtailment of equipment plan. Construction of the 4,500-Mark aircraft, the German Air Force wants only half size aircraft construction and the Zeppelin factories have announced intention to seek American capital to make possible its proposed building program.

and England to meet the demand for freight space which has increased considerably in the last few months. The company is likely to expect every day, it is reported, letters and telegrams from Holland leaving a large proportion of the aircraft.

British Columbia Airways is reported negotiating with the Washington Air Express, Inc., of Seattle, for an analysis of the two companies in the routes over which the two concerns have been operating practically coincide.

Sir Charles Wakefield, general manager of English continental system, has been for some time in South Africa a light plane. He has made similar visits to other clubs and has been the holder of Sir John Gold's rights to Australia and South Africa.



## MP-101 Acetylene Generator

To meet the demand for a stationary or portable acetylene generator adaptable for use in connection with either medium pressure or low pressure welding and cutting blow pipes, the General Acetylene Co., 30 East 42d St., New York City, has developed the First World Type MP-101 medium pressure acetylene generator. This generator has only been shown plans to place on a truck or wagon. Heavy page material has been used in the construction of the generator and heavier welding has been employed for joining the shell and flanges. All parts are either galvanized or chrome-plated as standard or according to the conditions in which they are likely to be subjected.



The First World Type MP-101 acetylene generator.

Plasma type motor feed control. This rotating spindle valve is self-cleaning, the carbide brushing off any dirt or lime deposit as it is spread. A pin on the stem above the valve prevents bridging of the carbide and consequent possible stoppage. There is nothing to get out of order in the carbide feed mechanism. The control is installed on a cast-iron housing.

Pressure adjustment is easily made by means of a thumb screw on the motor feed control. The operation is very sensitive. Photoaction during normal operation is less than 1 lb. per sq. in. and even under severe load the carbide feed is regulated to limit after-generation to a maximum increase of 1 lb. per sq. in.

Should, by any chance, both the motor brake and relief valves fail to function, the motor brake at about 18 lb. per sq. in. pressure, thereby stopping the carbide feed. Should a broken line or leak of any kind cause the pressure to drop to zero during operation, the motor stops and will not start again by hand.

Time required for shutting down the generator, flushing and refilling with carbide and water is about 15 min.

Relief valves are opened by raising an interference rod to vent the air-gas mixture after welding so that it is unnecessary to vent hose or pipe lines to the generator.

The feed valve once on is raised to vent the hopper when the generator is being moved. This valve also automatically raised, closing the hopper every time the carbide filling door is opened. The hopper and generator body are so designed that no water will enter the hopper in case the generator is knocked over.

The hydraulic back pressure valve and filter are connected to a stainless steel shell with a flange cover and cooling cooling, brazed-welded to the shell. The cover can be removed for replacing the filter. An interference and permit opening the water filling cock only while the relief valve is open. The Underwriters' Laboratories have permitted a double rating for the MP-101 generator making its rated capacity 60 or 80 per hour.

The carbide capacity of this generator is 30 lb. and the quarter size (1 1/4 in./12 in.) carbide is used. A special flange, which is held in place on top of the generator, is provided for filling the generator with carbide. The overall dimensions of the generator are: height, 60 in. and width, 34 in. The weight empty is 254 lb. With a full charge of carbide and water, the total weight is 623 lb. The weight gross for shipment is about 435 lb.

## Bundy Airplane Hangers

**THE BUNDY** hanger, offered by Carey Bundy, Pilot, Bunkley, Portland, Ore., is designed to house separately, individually and independently, any number of airplanes in a single structure. Plans are furnished



A Bundy hanger showing one or more bays tall of where

under floor or concrete are erected for construction of three hangers which are built of any desired combination of materials. The individual compartments in the hanger structures are "T" shaped, and are placed so that alternate compartments open on the same side of the structure. Light weight aluminum type doors are equipped.

A machine shop, store room, office or sleeping quarters can be built into the space above the tail

## Airplane Fueling Truck

A NEW type of service truck for refueling airplanes was recently approved by the U. S. Army Air Corps at Wright Field, Dayton, O. This truck is manufactured by the Columbus Steel Tank Company of Kansas City, Mo., which has received an order for four more of which will be used at Wright Field.

The truck which has been approved by the army is equipped with a 1000 gal. gasoline tank, 100 gal. of kerosene, 50 gal. water tank and an air compressor tank. The pumping apparatus and valves controlling the several tanks are located directly in front of the main gas tank.



One of the service trucks in use at the Toledo, Ohio Municipal Airport.

and directly back of the driver's cab. Power for forcing the fuel into the tanks of the airplane or dirigible is supplied from the truck engine.

Meters and glass gauges indicate the amount of fuel in the truck tanks and the plane can be refueled or equipped by reversing the pumps.

In addition to building this large type service truck the Columbus Steel Tank Company is also manufacturing a smaller truck for use on municipal airports. This type truck carries smaller tanks and is manufactured with both hand pumps and power pumps.

Realizing that fuel can be taken to airplanes, especially the large transport craft and dirigibles, much easier than the plane can be towed to a fixed fueling station, has prompted the Columbus organization to expand their business to take in this phase of the aviation industry.

## Gisholt Balancers

THE elimination of vibration in an airplane engine is not even greater importance than in an automobile engine. Airplane power plant manufacturers are following the lead of the automobile men with the result that the efficiency and life of many well known engines again has been greatly increased. Airplane engine manufacturers have found, however, that it is necessary to balance not only the crankshaft but also the propeller.

For propeller balancing the Gisholt Static Balancer has proved highly successful. This machine is really a fine laboratory testing device which has been developed for production purposes. Despite the large output of the machine and its simplicity of operation, it is sensitive to an imbalance of only 2 of an ounce inch. It will test this minute amount of imbalance in propellers or similar objects for which it is adapted.

For balancing crankshafts and other revolving parts the Gisholt Precision (Dynamic) Balancer has many features which appeal to airplane engine manufacturers. This machine also is sensitive to 2 of an ounce inch and conforms ease of operation with unvarying accuracy under production schedules.



## SIDE SLIPS

By Robert R. Osborn

Mr. M. A. G. of Holyoke, Mass., sends in a note describing the "Silver Cloud," a distinctive low-wing monoplane belonging to Captain James C. Pittsfield, Mass., an old English army flyer, who built the craft himself. It is also known as the "Down-Cause." It has a Chevrolet motor, a Ford radiator and other details which distinguish it easily from ordinary commercial ships. Its ceiling is said to be between five and six feet. Obviously designed for the great group of people who, when solicited for rides at the flying field, say they'd go up if they could keep out foot on the ground.

Mr. R. J. W. of Garden City, N. Y., sends in a clipping describing the purchase of "six Ryan-Monroey airplanes" by one man and asks for more information on this new company.

We don't recognize the company either but suppose it is the result of one of the many meetings going on all about us.

"BIRD'S DEEDS GET WELL EARNED REWARD" Headline in N. Y. Post.

We suppose that walking around an icy and undisturbed Antarctic landscape must be hard on one's feet.

We see by the papers that a routing welcome and street parade awaited after our American demonstration are planned for Major Segrave and The Golden Arrow car when they arrive in London. If the cause newspaper correspondents, afflicted with eye trouble, who covered the Florida speed trials are engaged in the parade we can expect something like the following:

"There was a parade from Waterloo station to the House of Parliament today. None of the spectators was able to turn his head fast enough to see what it was, but, according to reliable reports it consisted of Major Segrave in his Golden Arrow car accompanied by the Police Band and General Whalen."

"LADY HEATH FLIES Biplane 115 MILES AN HOUR" Headline.

And now what have these stories to say,—those who used to say the airplane would never amount to anything? We predict that speeds of 125 miles an hour will even be possible some day.

"Winnipeg, April 9 (A.P.) Charles (Synd) Johnson, the winner of various competitive air flights, successfully negotiated an outside loop in a biplane today. He was and to be the first flyer to perform the feat with an open biplane."

Possibly it was the first outside loop in a biplane carrying an electric stove.

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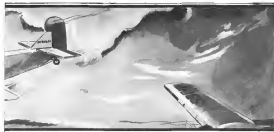
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This beacon can be furnished with a magnetic lamp changer and search lights which enable the operator to focus the beacon after he has passed over the main beam.

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WIND, VOL. 26, AVIATION

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Special wiring for all temperatures from -40 to +140, including: 1 discharge by motor, 1 standard starter for all types of aircraft whatever the power and number of motors may be.

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- (1) The main engine motor.
- (2) The main engine motor.
- (3) The main engine motor.
- (4) The main engine motor.
- (5) The main engine motor.
- (6) The main engine motor.
- (7) The main engine motor.
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Continental Model 18	18	18	18	18	18	18	18	18	18	18	18
Continental Model 20	20	20	20	20	20	20	20	20	20	20	20
Continental Model 22	22	22	22	22	22	22	22	22	22	22	22
Continental Model 24	24	24	24	24	24	24	24	24	24	24	24
Continental Model 26	26	26	26	26	26	26	26	26	26	26	26
Continental Model 28	28	28	28	28	28	28	28	28	28	28	28
Continental Model 30	30	30	30	30	30	30	30	30	30	30	30
Continental Model 32	32	32	32	32	32	32	32	32	32	32	32
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Continental Model 50	50	50	50	50	50	50	50	50	50	50	50
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# AERONAUTICAL ENGINEERING SECTION

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# The High Altitude Airplane

## The First of Two Articles Discussing Engineering Problems in the Design of an Airplane to Navigate in the Stratosphere

By B. V. KORVIN-KROUKOVSKY

PROGRESS of civilization is marked by an ever-increasing demand for rapid transportation, for the ability to cover greater distances in a shorter time. At first man could only walk and run, later he rode horses, then came sailing ships, then railroads, automobiles and finally airplanes. Now we are forced to look for means of increasing the speed of airplanes, particularly the speed over a long distance, of increasing the speed and at the same time increasing flying range. This is a difficult combination as the resistance of the air increases rapidly with speed, and the use of a powerful engine to overcome it results in a large gasoline consumption and a short flying range. Air resistance, the headwind force which gives lift to our wings and makes flight possible, now becomes our enemy, prevents us from increasing speed in a manner required by the progress of civilization. As we looked for means of reducing friction in developing railroads and automobiles, so now we must look for means of reducing air resistance.

Already great progress has been made in improving the shape of airplane parts, in streamlining to reduce the resistance, and it is expected that an airplane is being built in Germany consisting of but one large wing, with all parts creating possible resistance, such as the fuselage, completely eliminated. Nevertheless the wings, which we cannot eliminate, offer large resistance. We can reduce the resistance by reducing the size of wings, but there is a limit for such a procedure. If we have a large wing per square foot mass increases speed during take-off and landing, requires very large flying fields and is a source of danger. While we want to reach a great speed flying between cities and continents, we want to slow down to a low, safe speed while turning off and coming to a terminal. We want to find some way to reduce the air resistance, and to reach a great speed without decreasing the size of our wings and without increasing our landing speed. And fortunately the very nature of the airplane and of the atmosphere in which it flies points the way of doing it.

The atmosphere is not homogeneous, the air has the highest pressure and density at the ground level, and both the pressure and the density rapidly diminish with the altitude. The curves in Fig. 1 show that at the altitude of 22,000 ft the air is only half as dense as it is at sea level, and that at the altitude of 70,000 ft its density is reduced 25 times. The airplane, which by its very conception, is a machine capable of rising from the ground and of changing its altitude, can utilize the dense air at lower altitude for take-off and landing, and can utilize the reduced density of higher altitudes for reduction of air resistance and for increase in speed. In this paper we will consider how such a utilization is

possible at the present state of aeronautical engineering, and what obstacles we can derive therefrom. In this connection we will refrain from vast speculation as to what will be made possible by new inventions such as jet engines and jet propellers, and will consider only the power state of aeronautical engineering, and the improvements and devices which seem to be immediately available. There is no question that new inventions will follow as soon as their need is clearly established, but there is also no question that at early stages of the new development we will have to rely not on future, but on our present knowledge. It is essentially the purpose of

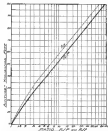


Fig. 1.—Pressure and density ratio variation with altitude

this paper to consider what we can do with the means available at present, with only such improvements as appear to be within easy reach.

The common way to handle an airplane on a cross-country flight is to take-off at a certain low speed, and at a certain large angle of incidence, to climb to a small altitude and then to reduce the angle of incidence by means of the elevator, thereby reducing the air resistance and increasing the speed. During the climb at a large angle of incidence comparatively small part of the engine power is needed for the resistance, and the excess of available power is used for the climb. In a horizontal flight, when the angle of incidence is reduced, the excess of power is used to overcome the extra air resistance pro-

duced by the increased speed. The maximum speed attainable in this way depends on the weight carried per horsepower and on the maximum speed of the airplane. The lower the maximum speed, the larger must be the wing area, the larger will be the air resistance, and the lower therefore will be the maximum speed. It has been shown by Delft that maximum speed of an airplane can be closely estimated by the formula:

$$(1) \quad V_{max} = K \sqrt[3]{P_{max}^{1/3} \frac{W}{S}} \sqrt{\frac{1}{\rho}}$$

where:

$P_{max}$  is the maximum speed in horizontal flight

$V_{max}$  is the maximum, or stalling speed

$W$  gross weight of airplane

$P$  engine horsepower

$\eta$  propeller efficiency

$K$  numerical coefficient

For modern commercial airplanes we can assume

$$K \sqrt[3]{\eta} = 28$$

and formula (1) then becomes

$$(2) \quad V_{max} = 28 P_{max}^{1/3} \sqrt{\frac{W}{S}} \sqrt{\frac{1}{\rho}}$$

In order to put the discussion on a concrete basis, let us assume a typical jet motor airplane with the following characteristics:

Gross weight	4,250 lb.
Engine power	425 hp
Wing area	380 sq ft.
Maximum lift coefficient	1.40
Minimum speed	55 mph

The maximum speed then works out as

$$V_{max} = 21 \times 55^{1/3} \times 380^{-1/2} = 141 \text{ mph}$$

In this paper we want to investigate a different method of handling an airplane, by which the initial large angle of incidence is not reduced, but is kept constant, and the increase of speed is obtained by means of decreased air density as the airplane climbs to a high altitude. At present we will assume that a supercharger or some similar device maintains the engine power constant regardless of the altitude, and later we will investigate and justify the validity of the assumption. At a given level our airplane can fly at a certain air speed  $V_a$ , slightly exceeding the minimum speed  $V_{min}$  and corresponding to the minimum power consumption  $P_a$ . Its wings develop the lift  $L$  equal to the weight of the airplane, and its resistance  $R$  is determined by its  $L/D$  ratio. The  $L/D$  ratio of an airplane depends only on its angle of incidence and does not depend appreciably on either speed or air density. In our case the angle of incidence remains constant during the climb, and therefore the  $L/D$  ratio also remains constant; the lift is always equal to the weight regardless of the altitude, and therefore remains also constant, from this it follows necessarily that the drag also must remain constant. As the airplane climbs, and the air becomes less dense with altitude, the speed of the airplane increases in order to maintain the same lift, and this increase of speed also serves to maintain the same drag. (Verify the mathematical expression we can write for the airplane

flying at a speed of minimum power consumption near the ground:

$$(3) \quad L = \frac{1}{2} \rho V_a^2 C_d A$$

and at any other altitude:

$$(4) \quad L = \frac{1}{2} \rho^0 V_a^0 C_d A$$

where:

$L$  = lift equal to weight of the airplane

$C_d$  = lift coefficient which is constant as long as the angle of incidence is constant

$A$  = wing area

$V_a$  = air speed at any altitude

$V_a^0$  = air speed at ground level

$\rho$  = air density at any altitude

$\rho^0$  = air density at ground level

Equating (3) and (4) and cancelling equal constant quantities we get:

$$\rho V_a^2 = \rho^0 V_a^0^2$$

or:

$$(5) \quad \frac{V_a}{V_a^0} = \left( \frac{\rho^0}{\rho} \right)^{1/2}$$

i.e., as the airplane climbs at a constant angle of incidence its speed increases in inverse proportion to the square root of air density. The power required to maintain the speed is obtained by multiplying the drag by the air speed, and as the drag in this case remains constant the power required is directly proportional to the air speed, and therefore is inversely proportional to square root of the air density, i.e.

$$(6) \quad \frac{P}{P_a} = \frac{P}{P_a^0} = \left( \frac{\rho^0}{\rho} \right)^{1/2}$$

Expressions (5) and (6) allow us to determine the climbing and the speed range of the airplane equipped with some kind of a supercharger maintaining constant engine power at any altitude, and flying at a constant angle of incidence. We observe that in this case the speed is directly proportional to the power, and not as the cube root of the power as was the case when the flight is at a constant altitude and varying angle of incidence was considered. This observation makes us expect that much higher speed may be attained by flying at a large angle of incidence at a great altitude, than flying at reduced angle of incidence over ground level.

It has been shown by E. P. Weir that minimum power required for the maintenance of the horizontal flight at ground level can be estimated by the formula:

$$(7) \quad P_a = \frac{R W \sqrt{L/A}}{L^{3/2}}$$

Substituting this value of  $P_a$  in expression (6), and assuming the propeller efficiency equal to 0.85, which is justifiable in view of the large loads necessary for the high altitude airplane, we get:

$$(8) \quad \frac{P}{P_a} = \frac{P}{P_a^0} = \frac{117}{\sqrt{L/A}}$$

substituting:

$$\frac{P}{A} = \frac{1}{2} \rho V_a^2 C_d$$

and assuming the value of  $C_L = 1.20$  as appropriate for the speed of minimum power consumption, evaluating  $\eta = 0.00237$  and introducing the correction coefficient of  $(52/15)^4$  so as to get the speed  $V$  in miles per hour we get

$$(9) \quad \frac{V}{V_0} = \left(\frac{P}{P_0}\right)^{1/3} = \frac{2,120}{P^{1/3} P_0}$$

or finally

$$(10) \quad V = \frac{2,120}{P^{1/3} P_0}$$

This is a very interesting and important result, showing that if the airplane is kept at a constant angle of incidence, and the lift is maintained by means of the climb in the altitude of greatly reduced air density the maximum speed obtainable is independent of the minimum speed.

The airplane with small wing area, high landing speed, and consequently high speed of minimum power consumption  $V_0$ , will not be able to climb very high, as shown by the value of the ratio  $P/P_0$  as obtained from (9) and its speed will be improved only a little. The airplane with large wing area will have very low speed  $V_0$  and  $V_0$  at the ground level, but it will be able to climb higher, and its speed will be improved much more. The maximum speed at the ceiling, however, will be the same in both cases, and will depend only on power loading  $W/P$ . We estimated above that a typical 425-hp airplane with a landing speed of 55 m.p.h. is able to attain the high speed of 141 m.p.h. at ground level. Observing that  $W/P = 10$  lb. per hp, we can estimate from (10) that were it kept at a large angle of incidence and made to climb to a high altitude in constant speed at the ceiling would be 212 m.p.h., the equivalent of 56 ft per sec over the ground level speed. Assuming the speed of minimum power consumption at sea level to be 40 m.p.h. corresponding to the landing speed of 55 m.p.h., we obtain from (9)

$$\frac{P}{P_0} = \left(\frac{2,120}{10 \times 40}\right)^3 = 42.5$$

The corresponding altitude is found by consulting the curves of Fig. 1 to be about 40,000 ft.

It is interesting to investigate the properties of the airplane which lead to the largest gain in speed at the altitude over that at ground level. Owing to formula (10) by (12) and denoting by  $V_0$  the maximum speed at the ground level, by  $V_c$  the maximum speed at the altitude, and by  $V_m$  the minimum speed at the ground level, we get:

$$(11) \quad \frac{V_c}{V_0} = \frac{10}{\left(\frac{W}{P} P_0\right)^{1/3}}$$

This shows that the gain made possible by high altitude flying increases with the decrease of the landing speed and with the decrease of the power loading. The tendency towards decreased power loading is the pre-dominating trend of aeronautical engineering, and every year we see more powerful engines and lower landings per horsepower. Every year, therefore, the possible gain due to high altitude flying will increase, until it becomes no longer to be expected. At the present time already a gain of 50 per cent can be expected, as has been shown by the example worked out earlier in this paper. The tendency towards increased landing speeds can be also observed in the present trend of

aeronautical engineering, but this can be considered only as a sacrifice in order to obtain higher maximum speed, and it is not necessary in the design of a high altitude airplane.

Next we will consider the effect of high altitude flying on the range, or distance which the airplane can cover with its supply of fuel. There are two distinct problems: one—the landing stage of several different airplanes and second—the constant range of several airplanes. The second landing airplane flies at about the constant angle of incidence at which its  $L/D$  ratio is the best. In this respect it is similar to the high altitude airplane. The difference lies in the fact that as fuel is consumed, and lighter weight leaves the excess of available power, this goes to a fairly small extent to be used for the climb, while in ordinary method of control the engine is throttled in order to economize the fuel. As in both cases the angle of incidence, and the  $L/D$  ratio remain constant, the range can be estimated by Bequest's equation:

$$(12) \quad \text{Range in miles} = 8635 \left(\frac{L}{D}\right) \left(\frac{P}{W}\right) \log_2 \left(\frac{W_0}{W_1}\right)$$

where

$W_0$  = actual gross weight in pounds.

$W_1$  = fuel gross weight in pounds.

$\eta$  = propeller efficiency at cruising speed.

$C$  = average specific fuel consumption.

It will be shown later that in a high altitude airplane, ground engines probably will be used and that the propeller always will operate at a constant and a very high efficiency, which we can estimate at least at 80 per cent. The engine runs nearly at full throttle and the specific fuel consumption can be taken at about 0.55 lb. per hp-hr. Then we get

$$\eta C = 0.80 \times 0.55 = 1.45$$

In an airplane operated at a low altitude and with varying throttle openings, the propeller usually not geared and its efficiency under normal about 70 per cent. The specific fuel consumption also varies from 0.55 lb. per hp-hr. at full throttle at the beginning of the flight, to 0.75 lb. per hp. at full throttle at the end of it, an average fuel consumption being about 0.65 lb. per hp. Using these figures we get

$$\eta C = 0.70 \times 0.65 = 1.00$$

Thus we see that while the same low power the second range of a common and a high altitude airplane, the details of the propeller efficiency and of fuel consumption give the latter the advantage of 1.45/1.00, or about 35 per cent improvement in the range.

In the above discussion we assumed the airplane set out to break the distance record, and fly at the best  $L/D$  ratio and at a low speed. In ordinary cases, on runs of from 300 to 600 mi., airplanes are not flown at such a low speed, and under "cruising" speed is usually about an arbitrary reduced speed adopted in order to save the engine. Usually it is between 75 per cent and 85 per cent of high speed. The range at such an arbitrary "cruising" speed is evidently proportional to the range at high speed, and the high altitude airplane showing 50 per cent improvement of speed for the same engine power, will show equal improvement in the same ratio, i.e., if it is considered practical to break the route of a low flying airplane into 300 mi. runs, these runs can be increased to 750 mi. for the high altitude airplane.

We have shown that flying at a very high altitude will

allow us to increase the speed some 50 per cent at the power loading used on present-day commercial airplanes, and that this gain over the maximum speed possible at sea level will increase further as the power loading will decrease with the development of more powerful and lighter engines. We have shown that this increase of speed will be accompanied by an increase of the range, thus reducing the necessary number of stops on long routes, and thereby increasing the average speed still more. Moreover we have shown that this increase of high speed will not be obtained at the sacrifice of landing speed, so it is at present, and that there will be no objection to keeping the landing speed as low as desired for safety, and in order to cut landing fields of moderate size. These are the basic advantages of the high altitude airplane, but they are not the only advantages. Improved safety, comfort of the passengers, ease of navigation, etc. are other important advantages. However, we will postpone the discussion of these, and will only address further proofs that high altitude flying

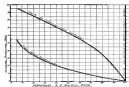


Fig. 1—Weight of air needed for combustion of 1 lb of gasoline.

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is feasible. So far we have spoken only about its advantages and merely assumed that it will be possible.

The possibility of high altitude flying depends on two basic requirements: first—the provision of an airframe in which the normal air density and temperature can be maintained despite the greatly reduced density and temperature of the surrounding atmosphere, and second—the arrangement of the power plant which will permit its operation in the extremely rarified and cold air. We will consider the question of the power plant first, as we must first prove the technical possibility of flying to high altitude and after that only need to consider the accommodation of the crew.

The horsepower of a plane, unsupercharged aviation engine decreases rapidly with altitude. The weight of the air entering the cylinders decreases as the density of the air decreases and so does the indicated horsepower. The friction losses, however, remain about the same regardless of the density, and these must be subtracted from the decreasing indicated horsepower. This makes the engine torque decrease more rapidly than the horsepower as the air decreases, i.e., more rapidly than the torque of the propeller decreases at any given rpm. The rpm of the engine thus decreases as the airplane climbs, and this decrease of rpm causes further loss of brake horsepower. Several engines have been tested in the altitude chamber of the Bureau of Standards at the low atmospheric pressure and temperature

closely simulating the conditions found at the altitudes. It has been found that the ratio of the power to the density can be approximated by the expression

$$HP = (P/P_0)^{1.54}$$

The variation of the horsepower with altitude calculated by this formula is plotted in Fig. 2. It will be observed that at the altitude of 20,000 ft. only about 41 per cent of the original horsepower is available and at 45,000 ft. only 10 per cent is available.

As the decrease of engine power occurs due to decrease in density it is evident that it can be prevented by compressing the air, and maintaining the ground pressure at the carburetor. A good deal of work has been done on the development of superchargers in all countries, and three distinct types have been evolved and brought to sufficient perfection for practical use. These types are the Roots Supercharger driven at maximum speed by a pair of gears from the engine crankshaft, the centrifugal supercharger driven at a very high speed by means of a gear train from the engine crankshaft, and the Turbo compressor, or the centrifugal supercharger driven by a turbine operated by exhaust gases. The first two types are identical in that the power needed to compress the air is taken from the crankshaft and should be subtracted from the brake horsepower of the engine causing the reduction of the horsepower, with the third, although not as large a net loss in case of the unsupercharged engine. The work needed to compress the air is subtractively expressed as:

$$(13) \quad HP_c = \frac{345}{33,000} \frac{1}{n-1} \left[ \left(\frac{P}{P_0}\right)^{\frac{n-1}{n}} - 1 \right]$$

where

$HP_c$  = Horsepower needed to drive the supercharger.

$n$  = The ratio of specific heat of gas at constant pressure to that at constant volume. For the air  $n = 1.405$ .

$K$  = Efficiency of the supercharger including driving gears.

$P$  = Pressure of the air at altitude in lb. per sq. in.

$P_0$  = Pressure of the air at sea level.

$V$  = Volume of the air at pressure  $P$  in cu ft.

The average aircraft engine consumes about 3 lb. of gasoline per horsepower per hour, and requires for its proper functioning 12 times as much air theoretically, and about 13 times as much air practically. Taking the density of the air in normal conditions as 0.002825 lb. per cu ft., the volume of the air required per horsepower per minute would be as:

$$0.5 \times 15/0.002825 \times 60 = 164 \text{ cu ft.}$$

The weight of the air needed for combustion of 1 lb of gasoline and for production of one brake horsepower does not change with altitude. As the pressure  $P$  decreases the volume  $V$  increases proportionally, and the product  $PV$  remains constant as long as temperature remains constant. The normal temperature at sea level is usually taken to 60 deg. F., or 288 deg. C. absolute and in a "Standard Atmosphere" it decreases at the rate of 1.98 deg. C. per thousand feet up to the altitude of about 35,000 ft., after which it remains constant at 238



into account we obtain finally the following values for the ratio of  $H_p/H_{p0}$ :

at 50,000 ft.	192 per cent
at 75,000 ft.	194 per cent
at 100,000 ft.	193 per cent

On basis of these figures we can say that in order to supply air at normal pressure to the engine at any altitude the turbo-supercharger must have the maximum overall efficiency of about 20 per cent. This figure appears to be quite modest and should not be difficult to attain.

The fact that at altitudes above 50,000 ft. or in the stratosphere, the miscellaneous gases gradually separate and tend to arrange themselves in order of their molecular weight, seems very to interesting speculation. The atmosphere contains not only nitrogen and oxygen, but a considerable amount of hydrogen, which it also about as the lower layers of the atmosphere, and predominantly in its upper layers. As the percentage of oxygen decreases and the percentage of hydrogen increases with altitude, the proportion is reached at the altitude of about 200,000 ft., where their content is just correct for combustion. Indeed, enormous explosions could have been caused by burning materials, were it not for the fact that combustion is impossible when the medium is so rarefied, as it is at the altitude of 200,000 ft. However, if this medium were compressed by a supercharger and by the piston of an engine, it would burn very well and would supply the power necessary for flight. Were an airplane built powerful enough to reach the altitude of 200,000 ft., it could remain there for indefinite length of time, and could cover any distance at a speed of some 3,000 mph. using atmosphere itself as a fuel. To be able to do so, however, it must possess about 40 times the maximum power it requires for horizontal flight at sea level.

So far we have considered only the question of the power required to drive the supercharger and of the net horsepower of the engine. We realized that we could see that simple power is available in the expansion of exhaust gases to supply the engine with air needed for full power consumption at all altitudes, provided that efficiency of the turbo-supercharger is not less than 20 per cent. Now we will consider whether it is possible or practical to provide the supercharger equipment of sufficient size and of so, what will be the probable weight of it. The existing superchargers built for use with a 400-hp engine, and capable of maintaining full pressure up to 20,000 ft. add about 150 lb. to weight of the engine plus the turbo-supercharger. It is reported that in only 2 lb. and the superchargers are built therefore as single-stage compressors with fairly high mechanical efficiency, and consume from 60 to 80 hp. Now it is estimated elsewhere in this paper that the excess of power which a modern airplane possesses leads us to expect that at an altitude of 75,000 ft. can be reached in a very near future. From formula (15) we estimate that a supercharger needed to supply a 400-hp engine with air at such an altitude will require 336 hp, i.e., about 4 times the power of the existing superchargers. It is quite certain that mechanical superchargers that reduce the use of any mechanical means all problems easier, while with increase of size all difficulties grow very rapidly. There is little doubt that powerful gas turbines will be built in the future, but as the near future, with which alone we are concerned in this paper,

it is not probable that 336-hp gas turbine can be built. Fortunately we do not have to do it, as we can get along quite well by using three or four superchargers of the existing size.

The Ramon supercharger is shown in Fig. 5, where (a) shows the inlet of exhaust gases, which must be contacted to the exhaust manifold of the engine, (b) shows the outlet for exhaust gases from the turbine, (c) shows the air inlet and (d) the air outlet leading to the intake manifold of the engine. It is quite possible to connect four turbines to that outlet of an air is associated with the inlet of the inlet, and the exhaust gases expand necessarily through the four turbines. The pressure ratio at the altitude of 75,000 ft. is 29.2, and the expansion per turbine is

$$\sqrt[4]{29.2} = 2.32$$

or, quite reasonable. If the connecting passages are made short and well insulated in order to prevent heat losses, expansion of the gas can be considered to be almost adiabatic, and formula (14) can be applied. In a similar way the first compressors will be contacted

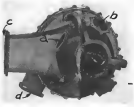


Fig. 5—The Ramon supercharger

in series, each one compressing the air 2.32 times, total compression being  $2.32^4$  or 29.2 times. Contrary to the turbine connection, the compressor connection should be constructed so as to have plenty of the cooling surface exposed to the air stream, so that heat generated by the compression on each stage can be dissipated.

When we speak of the supercharger of existing size, we mean that vital parts such as the turbine wheel, the shaft and the impeller are of the existing size and type. Of course there will be some difference in details, in particular in the number of stages through which the exhaust gases are directed into the turbine wheel. As the gases expand, the number or size of the nozzles in consecutive stages will have to be increased. In a similar way the size of inlet connections in the compressors will have to be increased, and probably the impeller wheel will have to be made wider in the later pressure stages. Finally it will be found expedient to increase the compression ratio per stage, reducing the number of stages to three, and equipping two turbo-compressors to handle the large volume of rarified air in the first compressor.

stage, supplying it to single turbo-compressors of the second and third stages. These, however, are details to be decided by the designer. All that remains in view is the fact that it is not only possible, but comparatively easy to provide superchargers of sufficient capacity.

The weight added by an early type turbo-compressor was 150 lb. for a 400-hp engine, which would make 600 lb. for the four stage turbo-supercharger discussed above. However, in combining four superchargers,

some economy in weight may be effected, and we must allow for the improvements made as the time goes on, so that we can maintain the weight of supercharger turbo-supercharging equipment designed to work at the altitude of 75,000 ft. at about one lb./hp., which is not prohibitive. In Part II, which will appear in the May 12 issue of the *Aeronautical Engineering Section*, of *AVIATION*, the problems of engine cooling, propeller airflow and cabin design of the high altitude airplane will be discussed.

## TECHNICAL REVIEWS

*N.A.C.A. Technical Report No. 305. The Gaseous Explosive Reaction, a Study of The Kinetics of Combustion Fuels, by F. W. Stromer*

This report deals with the results of a series of studies of the kinetics of gaseous explosive reaction where the fuel under observation, instead of being a simple gas, is a known mixture of simple gases. In the practical application of the gaseous explosive reaction as a source of power in the gas engine, the fuels employed are composite with characteristics that are apt to be due to the characteristics of their components and hence may be somewhat complex. The simpler problem that could be proposed in an investigation either of the thermodynamics or kinetics of the gaseous explosive reaction of a composite fuel would seem to be a separate study of the reaction characteristics of each component of the fuel and then a study of the reaction characteristics of the various known mixtures of these components bearing appropriate facts more and more complex. This is the order followed in the simple studies herein described.

*N.A.C.A. Technical Report No. 287, The Relation of Observed Airplane Performance to Standard Conditions, by Walter S. Dool*

This report shows how the actual performance of an airplane varies with air temperature when the pressure is held constant. This leads to comparatively simple methods of reducing observed data to standard conditions. The new methods which may be considered exact for all practical purposes, have been used by the Navy Department for about a year, with very satisfactory results.

The report also contains a brief historical review of the important papers which have been published on the subject of performance reduction, and traces the development of the standard atmosphere.

*N.A.C.A. Technical Report No. 306. Full Scale Wind-Tunnel Tests of a Series of Metal Propellers on a P.F.P. Airplane, by Fred E. Wirth*

An adjustable blade metal propeller was tested at five different angle settings forming a series varying in pitch. The propeller was mounted on a VE-7 airplane in the 30 ft. Propeller Research Tunnel of the National Advisory Committee for Aeronautics. The efficiencies were found to be from 4 to 7 per cent higher than those of standard wood propellers operating under the same conditions. The results are given in convenient form for use in selecting propellers for aircraft.

*N.A.C.A. Technical Report No. 298. Effect of Variation of Speed and Span of Airplane on Rolling and Yawing Moments in Level Flight, by R. H. Hoell and D. H. Strader*

This report presents the results of an investigation of the rolling and yawing moments due to adverse of various clouds and spins on two airfoils having the Clark Y and U.S.A. 27 wing sections. Some attention is devoted to a study of the effect of scale on rolling and yawing moments and to the effect of airfoil rounding the wing tip.

The results apply to level flight with the wing chord set at an angle of attack of 4-5 deg. and in conditions of zero pitch rate zero yaw, and zero roll of the airplane. It is planned here to extend the investigation to other attitudes for monoplane and biplane configurations.

The work was conducted in the 10-1 wind tunnel of the Bureau of Standards on models of 10 in. span and 10 in. chord.

*Department of Commerce, Bureau of Standards Circular No. 360, Summarizing of Airplane Data, by J. L. Cavalieri and H. F. Lipsett*

The article contains a report of the work done in determining the structure which will give the maximum amount of sound radiation in an airplane cabin for a constant weight.

Various metal structures were tested at frequencies varying from 100 to 1,100 cycles per second to determine the best structure available within the allowable limit of weight.

A test flight was made in a treated cabin to determine how satisfactory the structure was under operating conditions. It was found that the noise in the cabin was appreciably less severe so that in the interior of a rail way coach is noise.

*N.A.C.A. Technical Report No. 302. Full Scale Tests on a Thin Metal Propeller at Various Tip Speeds, by Fred E. Wirth*

This report describes an investigation made in order to determine the effect of tip speed on the characteristics of a thin-bladed metal propeller. The propeller was mounted on a VE-7 airplane with a 180-hp. E-2 engine and tested in the 30 ft. propeller research tunnel of the National Advisory Committee for Aeronautics. It was found that the effect of tip speed on the propulsive efficiency was negligible within the range of the tests which was from 600 to 1,000 ft. per sec. (about 0.5 to 0.9 the velocity of sound in air).

# Relative Lift Distribution In Any Biplane

By L. V. KIBBER\*

**A** KNOWLEDGE of the distribution between the upper and lower wings of the total lift of a biplane is necessary in the solution of two airplane design problems, one of an aerodynamic nature and the other of an aerodynamic nature. In the structural analysis of a biplane airframe, the relative loading on the wings has a marked influence on the stresses in the spars. In the computation of the performance of a biplane, the induced drag of the airframe is a main factor. Prandtl<sup>1</sup> has shown that it depends on the lift distribution, and gave particular solutions for maximum induced drag on the assumption that lift distribution is proportional to area distribution. A recent report<sup>2</sup> gave a general solution of Prandtl's equation for any lift distribution.

Numerous wind-tunnel tests of lift distributions have been made on models of equal-span, equal-chord biplanes with various combinations of gap/chord and stagger. With this empirical data as a basis, theoretical considerations permit a solution for biplanes with unequal spans and unequal chords, and with or without stagger. In the present report, charts are derived which give the solutions of the general cases. Three regimes of the polar are treated, that is, 25, 50, and 90 per cent of  $c_{L/D}$  maximum, important for the low incidence stress analysis case, for the determination of a representative value of the induced drag per unit chord, and, for the high lift, high incidence stress analysis case, respectively. The relative loading on the upper and lower wings,  $\epsilon$ , is first given as a function form for various values of  $\epsilon$ , the ratio of chord, and for  $\epsilon$ , the ratio of span, equal to unity. Then a formula is derived which allows the introduction of  $\epsilon$  into this unity. Finally, a weighted solution gives a correction to  $\epsilon$  for the effect of dissymmetry. The variables are the ratio of the gap to the chord of the upper wing,  $r$ ,  $\epsilon$ , stagger, dissymmetry, and  $\epsilon$ , the proportion of the total lift contributed by the upper wing. The effect of variation in gap/chord on the basic form of  $c_{L/D}$  is readily apparent, and the omission of this variable greatly simplifies the whole problem.

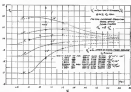
## The Basic Data

In the past eight years, quite a number of isolated and limited tests have been made of the lift distribution in equal-span, equal-chord biplanes.<sup>3</sup> Lately, the NACA published the results of a more systematic and comprehensive series of tests on an RAE 15 biplane in which stagger and gap/chord were varied in rather wide limits. These data are plotted in Fig. 1, together with some of the earlier data, in the basic form of  $c_{L/D}$ .

\*Nomenclature Professor of Applied Aerodynamics, University of Michigan, Ann Arbor, Michigan. Present address: National Bureau of Standards, Washington, D.C. 20535.  
†NACA Tech. Rept. No. 115.  
‡NACA Technical Staff, "Biplanes", NACA Rept. No. 107.  
§NACA Tech. Rept. No. 115.  
||See footnote on Page 149.

is function of the ratio  $G/r_0$  of the gap to the chord of the upper wing, and of the stagger. At any value of  $\epsilon$  of the biplane,  $\epsilon'$  is equal to the lift coefficient of the upper divided by the lift coefficient of the biplane.

The ratio  $\epsilon'$  was to be fair the test data, which show a wide dispersion. The NACA data were given more weight than the older aerodynamic tests on smaller models at lower test velocities. Two theoretical considerations simplify the charting. First, at larger values of  $G/r_0$ , the  $\epsilon'$  values went toward unity, and then for all values of stagger. Second, when the gap is zero, the two wings coincide and  $\epsilon'$  is again unity. That



this is true can also be deduced when the gap is taken smaller and smaller until the lower wing passes through the upper, when the gap may be considered negative. The upper wing now becomes the lower, and we must plot values of  $\epsilon' = 2 - \epsilon''$  (as in Fig. 1) on the left-hand side of  $G/r_0 = \text{zero}$ . The only possible continuous function is the one which passes through  $\epsilon' = \text{zero}$  at  $G/r_0 = \text{zero}$ .

With these considerations in mind and by careful cross-plotting of the data for values of  $G/r_0$  or constant, the curves of Fig. 1 were drawn as probably most representative of the actual conditions in an RAE 15 biplane. Figs. 2 and 3 incorporate similar data at other values of  $\epsilon/r_0$ .

Values of  $\epsilon'$  derived from the data on the Air Corps Handbook for Designers<sup>4</sup> have been plotted in Figs. 1 and 3 for comparison with the new data and indicate, first, that they are just as inaccurate as high as at low values of  $\epsilon$  and, second, that the old experimental data had been miserably faked.

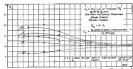
It is regretted that more extensive tests of the USAF section No. 5 were not made, as the data observed at  $G/r_0 = 0.9$  not only do not show the RAE 15 data,

\*From chart Fig. 3, Bulletin 14, Aeronautical Research, Department of Commerce, U. S. Government.

but the great change in  $\epsilon'$  from stagger +30 deg. to zero stagger in conjunction with the change from zero stagger to stagger -30 deg. appears highly improbable. Just why the relative loading at a certain  $\epsilon$  value should be expected to show any appreciable variation with the airfoil thickness or camber is not clear.

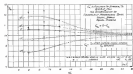
## The Relative Loading

The relative loading is designated by  $\epsilon$  and is the ratio of the lift coefficient of the upper to the lift coefficient of



the lower wing. It is used directly in the stress analysis of wings. The present data are based on older tests on equal-span, equal-chord biplanes, but have been indiscriminately used for unequal chords and spans and for both the high and low incidence conditions. In this report empirical-theoretical data are given for 25 per cent and 90 per cent  $c_{L/D}$  for stress analysis purposes.

The relative loading can also be used to determine  $L/D$ , the lift of the upper wing divided by the total lift. This ratio has been determined as  $\epsilon'$  in ACIC 8607. Strictly speaking, there is a different value of  $\epsilon$  all along the polar, as that the induced drag curve is the envelope of a family of parabolas. The induced drag is not an important factor at low  $\epsilon$  values, nor is it at high  $\epsilon$  values, which are outside the practical structural flight range. However, at medium  $\epsilon$  values, where climbing occurs the induced drag is a main factor, so that this report gives data on  $\epsilon$  and, hence on  $\epsilon'$ , for  $\epsilon$  equal to 50 per cent  $c_{L/D}$ . It is intended that the parabola through this point be substituted for the above mentioned



envelope curve, that is, that  $\epsilon$  in ACIC 8607 be computed from values in this report of  $\epsilon$  at 50 per cent  $c_{L/D}$ .

## 2. Equal Span, Equal Chords ( $r = 1$ , $\epsilon \geq 1$ )

In this simplest case, where the areas of the upper and lower wings are equal,  $\epsilon$  may be found from the basic

data. The left of the biplane is the mean of the lifts of the two wings.

Since

$$A_1 = A_2 = \frac{A}{2} \quad (1)$$

In a conventional biplane having equal chords and equal spans,  $A_1$  will be greater than  $A_2$  owing to the portion of the wing cut by the fuselage. The problem is greatly simplified by neglecting this effect, which is considered negligible.

By definition,

$$\epsilon' = \frac{c_{L1}}{c_L} = \frac{c_{L1}}{c_{L1} + c_{L2}} \quad (2)$$

Then,

$$\epsilon' = 2 - \epsilon' \quad (3)$$

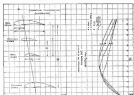
and

$$\epsilon = \frac{c_{L1}}{c_{L2}} = \frac{c_{L1}}{c_{L1}/(2 - \epsilon')} \quad (4)$$

$\epsilon'$  is taken from Figs. 1, 2, and 3;  $\epsilon$  is computed from Equation (3) and plotted in Figs. 1, 2, and 3.

## 2. Equal Span, Unequal Chords ( $r = 1$ , $\epsilon \geq 1$ )

A simple theoretical consideration permits an approximate determination of this case. In Fig. 29 it is ap-



proved such a biplane in which  $\epsilon/r_0 = \epsilon = 0.5$ . The problem is to estimate the effect of the lower on the upper wing and of the upper on the lower wing. Since the lift is directly proportional to both area and circulation and since the circulation velocity varies inversely with the distance from the disturbing wing, the same disturbance velocities result at the upper wing from the circulation around a fictitious lower wing with twice the chord (hence, twice the area of the real lower) and twice as far away from the upper as is the real lower. So, if we want to find  $\epsilon'$  for  $G/r_0 = 1.0$  where  $\epsilon = 0.5$ , we must take  $\epsilon'$  from Fig. 1<sup>†</sup> at  $G/r_0 = 2.0$  since we now have to consider a biplane of equal span and equal chord but with twice the gap. Similarly, the effect of the upper on the lower can be approximated by placing a fictitious upper wing with half the chord of the real upper at half the real gap away from the lower. To find  $\epsilon'$  =  $2 - \epsilon'$ , we take  $\epsilon'$  from Fig. 1<sup>†</sup>, but at the original  $G/r_0$ . In general, we take

†† See Table from Figures 2 or 3, on the next to next page.



Case III—Unequal span, unequal chords, no drag.

$$\frac{r}{c} > 1 \\ \frac{r}{c} < 1$$

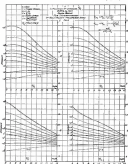
1. Determine  $r$  for  $r = 1$  from Figs. 4 to 27, as the case may be.
2. Determine  $c_1/c_2$  from Figs. 1, 2, or 3, as required.
3. Determine  $r$  for  $r \neq 1$  from the following relations:

$$a. \quad r(r < 1) = r \cdot r(r=1) + \frac{1-r}{2-c_1/c_2}$$

$$b. \quad r(r > 1) = r \cdot r(r=1) \left[ 1 - \frac{1-r}{c_1/c_2 + r} \right]$$

Case IV—Case I, II, or III with drag.

1. Determine  $r$  for no drag as above.
2. Calculate  $a$  for no drag from  $a = \frac{c_1 A_1}{c_2 A_2}$ .
3. Determine  $\frac{a}{r}$  from Fig. 28.
4. Calculate  $a_d$  from Equation (3) (first approximation).
5. Recalculate  $a$  from  $a = \frac{a_d}{c_1 A_1 + A_2}$ .
6. Determine  $\frac{a}{r}$  from Fig. 28.
7. Calculate  $a_d$  from Equation 6 (second approximation).

(Use use in charts of A.C.I.C. 602, the second approximation for  $a$  must be used.)

## Examples

The lift distribution in representative biplane outlines is solved in the following according to the formulas and charts of this paper:

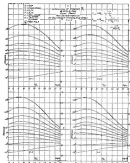
Case I—Equal span, equal chords, no drag.

Span—Upper wing	= 40 ft. ( $b_1$ )
Span—Lower wing	= 40 ft. ( $b_2$ )
Chord—Upper wing	= 60 in. ( $c_1$ )
Chord—Lower wing	= 60 in. ( $c_2$ )
Gap	= 60 in. ( $G$ )
Stagger	= +15 deg.
$r = b_1/b_2$	= 1
$a = c_1/c_2$	= 1
$G/c_1 = 60/60$	= 1

From Fig. 5,  $r = 1.185$  (Low incidence).From Fig. 13,  $a = 1.221$  (High incidence).From Fig. 21,  $r = 1.210$  (For determination of equivalent monoplane span).

Case II—Equal span, unequal chords, no drag.

Span—Upper wing	= 40 ft. ( $b_1$ )
Span—Lower wing	= 40 ft. ( $b_2$ )
Chord—Upper wing	= 60 in. ( $c_1$ )
Chord—Lower wing	= 30 in. ( $c_2$ )
Gap	= 60 in. ( $G$ )
Stagger	= +15 deg.
$r = b_1/b_2$	= 1
$a = c_1/c_2 = 30/60$	= 0.5
$G/c_1 = 60/60$	= 1

From Fig. 10,  $r = 1.123$  (For low incidence).From Fig. 28,  $a = 1.150$  (For high incidence).From Fig. 36,  $r = 1.132$  (For determination of equivalent monoplane span).

Case III—Unequal span, unequal chords, no drag.

Span—Upper wing	= 40 ft. ( $b_1$ )
Span—Lower wing	= 20 ft. ( $b_2$ )
Chord—Upper wing	= 60 in. ( $c_1$ )
Chord—Lower wing	= 30 in. ( $c_2$ )
Gap	= 60 in. ( $G$ )
Stagger	= +15 deg.
$r = b_1/b_2 = 20/40$	= 0.5
$a = c_1/c_2 = 30/60$	= 0.5
$G/c_1 = 60/60$	= 1

From Fig. 10,  $r(r=1) = 1.121$ 

(For low incidence).

From Fig. 1,  $a_1 = 1.090$ 

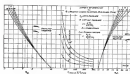
(For equal chords and equal span)

$$r(r < 1) \cdot r(r=1) + \frac{1-r}{2-c_1/c_2}$$

$$= 0.5 \times 1.121 + \frac{1-0.5}{2-1.090}$$

$$= 0.560 + 0.916$$

$$= 1.476$$

 $r(r < 1) = 1.129$  for the tip incidence condition.The value of  $r$  for the high incidence condition and for the determination of the equivalent monoplane span

may be determined in the same manner by the use of the proper charts.

Case IV—Case I, II, or III with drag.

Area upper wing	= 200 sq ft. ( $A_1$ )
Area lower wing	= 45 sq ft. ( $A_2$ ) (includes 20 per cent fuselage and 25 per cent nacelle)

Span upper wing = 40 ft. ( $b_1$ )Span lower wing = 20 ft. ( $b_2$ )Chord upper wing = 60 in. ( $c_1$ )Chord lower wing = 30 in. ( $c_2$ )

## TECHNICAL PUBLICATIONS RECEIVED

N.A.C.A. Technical Note No. 304, Correlation of Wind-tunnel and Full-Scale Results of a smaller aspect ratio, by Harry S. Ransden, Bureau of Standards.

N.A.C.A. Technical Memorandum No. 908, Impact Water and Detonation, by R. Becker. Part I. From *Zeitschrift für Physik*, Volume VIII.

N.A.C.A. Technical Memorandum No. 504, *Mathematical Calculations of Airplanes*, by H. Schlichting. From the 1932 yearbook of the *Wissenschaftliche Gesellschaft für Luftfahrt*.

Gap	= 60 in. ( $G$ )
Stagger	= +15 deg.
Drag	= ± 5 deg.

The above characteristics are identical with Case III, dragless coupled.

 $r = 1.199$  (no drag) (from Case III, low incidence) $a = \frac{c_1 A_1}{c_2 A_2}$  $a = \frac{1.199 \times 200}{(1.090 \times 45) + 45} = \frac{221.8}{221.8 + 45} = \frac{221.8}{266.8}$  $a = 0.832$  $\frac{a}{r} = 1.46$  (from Fig. 28 at  $a = 0.008 = 25$  per cent  $a_{max}$ ) $r = 1.199 \times 1.46 = 1.639$  (first approximation) $a = \frac{1.199}{c_1 A_1 + A_2} = \frac{1.199}{1.62 \times 3.6 + 45} = \frac{324}{324 + 45} = \frac{324}{369} = 0.878$  $\frac{a}{r} = 1.46$  (from Fig. 28 at  $a = 0.008, 25$  per cent  $a_{max}$ ) $r = 1.46 \times 1.199 = 1.642$  (second approximation)

The value of  $a_d$  for the high incidence condition and for the determination of the equivalent monoplane span may be determined in the same manner, using the proper value of  $r$ , and the  $a$  in Fig. 28.

## Summary

With experimental data on typical biplane outlines as basis, the empirical-theoretical method of this paper permits the general solution for the lift distribution in any biplane. The determination of the relative loading  $r$  for stress analysis, made is indicated, and the effect of several lift distributions on induced drag is shown. The data of this paper in conjunction with the charts of a previous report, A.C.I.C. 602, on "The Induced Drag of Any Biplane." The introduction of the variable  $c_1/c_2$  into the charts is held to be necessary and undesirable. Sweepback, dihedral, and taper may be eliminated from the problem by dealing with mean aerodynamic chords of the two wings and with a mean value of the gap. It is believed that the data of this paper, although based solely on N.A.C.A. 15 biplane tests, may be considered valid for biplanes with airfoils of other thicknesses and numbers, at least until further systematic tests have been made which show differences due to airfoil section profile than the probable error of any one measurement.

Department of Commerce, Bureau of Standards, Research Paper No. 48 Transmission of Sound Through Wall and Floor Structures, by F. L. Christy, associate physicist and W. F. Snyder, junior physicist.

Division of Shipfitted Practice, Department of Commerce, results and benefits of applying Shipfitted Practice in Curlew Breeds and Beach Sheeps, by H. P. DeWolf, Research Unit.

Royal Aeronautical Society, *Aircraft-Body Interference*. A paper by C. N. H. Leach, M. A.





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